Rotor 35

- Français
- English

Downloadable files
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Open access

Git project

Original model

Rotor 35 is part of a research program to study a advanced-core compressor design with a high compression ratio (20:1). It is therefore the first stage rotor of this eight stage transonic compressor. Of these eight stages, the first four have been designed and tested : rotors 35, 36, 37 and 38. For more information, here is a link to a report from NASA.

- Original technical report [1]:

```
@TechReport{reid1982design,
  author          = {Reid, L. and Moore, R. D.},
  date            = {1982},
  institution     = {NASA Lewis Research Center Cleveland, OH, United
States},
  title           = {Performance of single-stage axial-flow transonic
compressor with rotor and stator aspect ratios of 1.63 and 1.78, respectively,
and with design pressure ratio of 1.82},
  number          = {NASA-TP-1974},
  url             = {https://ntrs.nasa.gov/citations/19820011348},
}
```

- Picture :

Fig. 1 https://catalog.archives.gov/id/17466807

```
@Misc{brown1977records,
author    = {Brown, M.},
title     = {Rotor 35 - Rotor 35 - Stator 35 in casing. {R}ecords of the
{N}ational {A}eronautics and {S}pace {A}dministration, 1903 - 2006.
{P}hotographs relating to agency activities, facilities and personnel,
1977 - 2013},
url     = {https://catalog.archives.gov/id/17466807}}
```

Useful documents

- downloadable models (Git project)
- PDF of the NASA report :
  rotor35.pdf
- CSV file of the blade geometry :
  rotor35_original.csv
  , usable as input of OpenMCAD[2] to generate reference blade models.

Reference blade

The **reference blade** is defined with multiple-circular arc profiles[3] given in the original NASA report[1]. Corresponding models are computed with the open-source code OpenMCAD[2].

Geometry

[The geometry of rotor 35 is described in the original NASA report](#) by the following tables. The length are in centimeters and the angles in degrees.

(a) For rotor 35

```
      PERCENT    RADII          BLADE ANGLES        DELTA    CONE
RP     SPAN    RI    RO       KIC    KTC    KOC      INC    ANGLE
T1P     0.  25.248 24.511   62.55  62.99  53.21    2.09  -15.764
 1      5.  24.916 24.221   61.52  61.84  52.53    2.41  -14.327
 2     10.  24.571 23.931   60.55  60.74  51.87    2.72  -12.780
 3     15.  24.224 23.642   59.80  59.85  51.23    2.96  -11.326
 4     30.  23.163 22.772   58.34  57.74  48.54    3.41   -7.137
 5     50.  21.726 21.613   56.16  54.31  44.26    4.21   -1.890
 6     70.  20.221 20.454   53.70  49.53  39.16    5.51    3.545
 7     85.  19.019 19.584   52.28  47.30  33.31    6.56    8.150
 8     90.  18.596 19.294   52.00  46.85  30.96    6.86    9.887
 9     95.  18.158 19.005   51.82  46.50  28.36    7.18   11.763
HUB   100.  17.780 18.715   51.69  46.24  25.70    7.46   12.787


      BLADE THICKNESSES           AXIAL DIMENSIONS
RP    TI     TM     TO      ZI     ZMC    ZTC     ZO
T1P  .025   .175   .025    .698   2.410  2.379  3.308
 1   .027   .187   .027    .635   2.531  2.345  3.354
 2   .028   .199   .028    .576   2.313  2.301  3.398
 3   .029   .212   .029    .529   2.268  2.242  3.438
 4   .032   .252   .032    .417   2.188  2.051  3.542
 5   .037   .305   .038    .280   2.133  1.896  3.701
 6   .042   .361   .043    .129   2.045  1.749  3.884
 7   .047   .408   .047    .058   1.992  1.715  4.007
 8   .048   .425   .049    .037   1.967  1.646  4.046
 9   .050   .443   .050    .017   1.940  1.579  4.082
HUB  .051   .458   .051    .000   1.915  1.520  4.118
```

Aerodynamic design

|	unit	values
pressure ratio	[-]	1.82
mass flow	[kg/s]	20.2
tip speed	[m/s]	455
tip solidity	[-]	1.3
aspect ratio	[-]	1.19
number of blades	[-]	36
nominal rotation speed ω_n	[rad/s]	1800

Material properties

Rotor 35 is made of a 200-grade maraging steel[4], but the exact material properties are not provided in the NASA report. The following properties are considered :

	unité	valeurs
alloy	[-]	18-Ni-200-maraging
Young's modulus	[GPa]	180
density	[kg/m3]	8000
Poisson's ratio	[-]	0.3
yield stress	[GPa]	1.38

CAD model

The CAD model is computed with the open source code OpenMCAD[2].
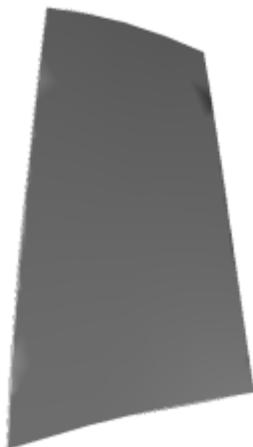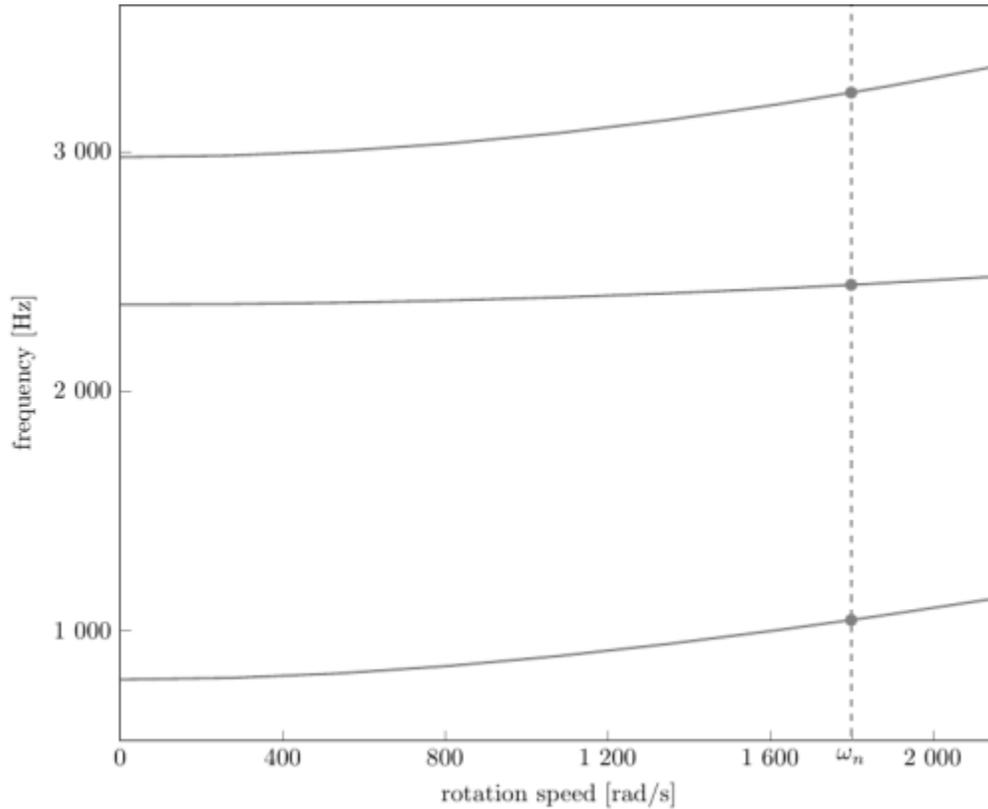

pressure side

suction side

Natural frequencies

First three natural frequencies (with clamped root) for the mesh computed with OpenMCAD[2]:

Mode	Type	Natural angular frequency (rad/sec)	Natural frequency (Hz)
1	1B	5000.55	795.863
2	1T	14833.91	2360.89
3	2B	18699.83	2976.17

Campbell diagram

Evolution of the natural frequencies of the first 3 vibration modes, as a function of rotation speed, for the mesh computed with OpenMCAD[2]:

Campbell diagram computed with a linear centrifugal preload, with clamped root (nominal rotation speed ω_n = 1 800 rad/s)

- graph (.pdf)
- Campbell data (.csv)

Initial blade

The **initial blade** is defined with in-house LAVA parameters[5] computed from the reference blade CAD model. The initial blade is usually used as starting point for an optimization process. Its geometry is similar to the one of the reference blade.

Natural frequencies

First three natural frequencies (with clamped root)

- from the whole mesh:

Mode	Type	Natural angular frequency (rad/sec)	Natural frequency (Hz)
1	1B	5037.31	801.712
2	1T	14910.06	2373.01
3	2B	18900.76	3008.15

- from the reduced order model:

Mode	Type	Natural angular frequency (rad/sec)	Natural frequency (Hz)
1	1B	5037.84	801.797

Mode	Type	Natural angular frequency (rad/sec)	Natural frequency (Hz)
2	1T	14917.41	2374.18
3	2B	18926.96	3012.32

Campbell diagram

Comparison of the evolution of the natural frequencies of the first 3 vibration modes, as a function of rotation speed for the initial and the reference blade:
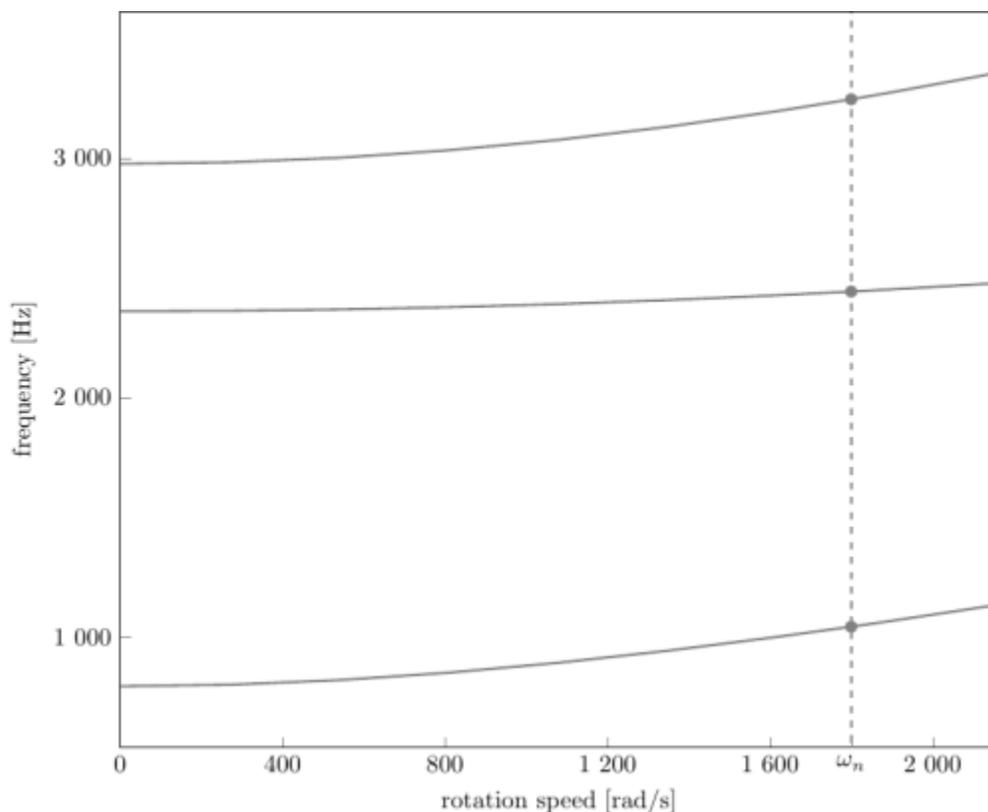
Campbell diagram computed with a linear centrifugal preload, with clamped root (nominal speed ω_n = 1 800 rad/s),

initial blade (orange), reference blade (gray)

- graph (.pdf)
- Campbell data (.csv)

Fichiers téléchargeables
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Libre accès

lien vers le projet Git

Modèle original

Le rotor 35 appartient à un programme de recherche visant à étudier une conception de compresseur possédant un grand taux de compression (20:1). Il est donc le rotor du premier étage de ce compresseur transsonique de huit étages. Parmi ces huit étages, les quatre premiers ont été conçus et testés, ils correspondent aux rotors 35, 36, 37 et 38. Pour plus d'information, voici un lien vers un rapport de la NASA.

- Rapport technique original [6]:

```
@TechReport{reid1982design,
  author          = {Reid, L. and Moore, R. D.},
  date            = {1982},
  institution     = {NASA Lewis Research Center Cleveland, OH, United
States},
  title           = {Performance of single-stage axial-flow transonic
compressor with rotor and stator aspect ratios of 1.63 and 1.78,
respectively, and with design pressure ratio of 1.82},
  number          = {NASA-TP-1974},
  url             = {https://ntrs.nasa.gov/citations/19820011348},
}
```

- Photographie :

Fig. 1 https://catalog.archives.gov/id/17466807

- ```
  @Misc{brown1977records,
  author  = {Brown, M.},
  title   = {Rotor 35 - Rotor 35 - Stator 35 in casing. {R}ecords of the
  {N}ational {A}eronautics and {S}pace {A}dministration, 1903 - 2006.
  {P}hotographs relating to agency activities, facilities and personnel,
  1977 - 2013},
  url     = {https://catalog.archives.gov/id/17466807}}
  ```

Documents utiles

- modèles téléchargeables (lien vers projet Git)
- PDF du rapport de la NASA :

  rotor35.pdf

- Fichier CSV de la géométrie :

  rotor35_original.csv

  , utilisable en entrée de OpenMCAD[2] pour générer l'aube de référence

Aube de référence

L'**aube de référence** est définie par des profils de type arcs circulaires multiples[3], donnés dans le rapport technique original de la NASA[6]. Les modèles associés sont obtenus avec le code en libre accès OpenMCAD[2].

Géométrie

La géométrie du rotor 35 est décrite dans le rapport d'origine de la NASA par les tableaux suivants. Les grandeurs sont en centimètres et en degrés.

(a) For rotor 35

```
        PERCENT    RADII         BLADE ANGLES      DELTA   CONE
RP      SPAN   RI    RO        KIC    KTC    KOC    INC   ANGLE
TIP     0.   25.248 24.511    62.55  62.99  53.21  2.09  -15.764
1       5.   24.916 24.221    61.52  61.84  52.53  2.41  -14.327
2      10.   24.571 23.931    60.55  60.74  51.87  2.72  -12.780
3      15.   24.224 23.642    59.80  59.85  51.23  2.96  -11.326
4      30.   23.163 22.772    58.34  57.74  48.54  3.41   -7.137
5      50.   21.726 21.613    56.16  54.31  44.26  4.21   -1.890
6      70.   20.221 20.454    53.70  49.53  39.16  5.51    3.545
7      85.   19.019 19.584    52.28  47.30  33.31  6.56    8.150
8      90.   18.596 19.294    52.00  46.85  30.96  6.86    9.887
9      95.   18.158 19.005    51.82  46.50  28.36  7.18   11.763
HUB   100.   17.780 18.715    51.69  46.24  25.70  7.46   12.787


        BLADE THICKNESSES        AXIAL DIMENSIONS
RP      TI    TM    TO       ZI    ZMC    ZTC    ZO
TIP    .025  .175  .025    .698  2.410  2.379  3.308
1      .027  .187  .027    .635  2.361  2.345  3.354
2      .028  .199  .028    .576  2.313  2.301  3.398
3      .029  .212  .029    .529  2.268  2.242  3.438
4      .032  .252  .032    .417  2.188  2.051  3.542
5      .037  .305  .038    .280  2.133  1.896  3.701
6      .042  .361  .043    .129  2.045  1.749  3.884
7      .047  .408  .047    .058  1.992  1.715  4.007
8      .048  .425  .049    .037  1.967  1.646  4.046
9      .050  .443  .050    .017  1.940  1.579  4.082
HUB    .051  .458  .051    .000  1.915  1.520  4.118
```

Caractéristiques aérodynamiques

	unités	valeurs
taux de compression	[-]	1,82
débit massique	[kg/s]	20,2
vitesse en tête	[m/s]	455
solidité en tête	[-]	1,3
allongement	[-]	1,19
nombre d'aubes	[-]	36
vitesse de rotation nominale ω_n	[rad/s]	1800

Propriétés matériau

Le matériau du rotor 35 est un alliage à base de nickel : un acier maraging de grade 200[4], mais ses caractéristiques ne sont pas fournies dans le raport de la NASA. Les propriétés considérées sont :

	unité	valeurs
alliage	[-]	18-Ni-200-maraging
module d'Young	[GPa]	180
masse volumique	[kg/m3]	8000
coefficient de Poisson	[-]	0,3
limite élastique	[GPa]	1,38

Modèle CAO

Le modèle CAO est obtenu avec OpenMCAD[2].

intrados

extrados

Fréquences propres

Fréquences des trois premiers modes (noeuds du pied d'aube encastrés) pour le maillage obtenu avec OpenMCAD[2] :

Mode	Type	Pulsation propre (rad/sec)	Fréquence propre (Hz)
1	1F	1018,5	162,1
2	1T	3542,5	563,8
3	2F	6094,7	970,0

Diagramme de Campbell

Évolution des fréquences propres des 3 premiers modes, en fonction de la vitesse de rotation, pour le maillage obtenu avec OpenMCAD[2]:
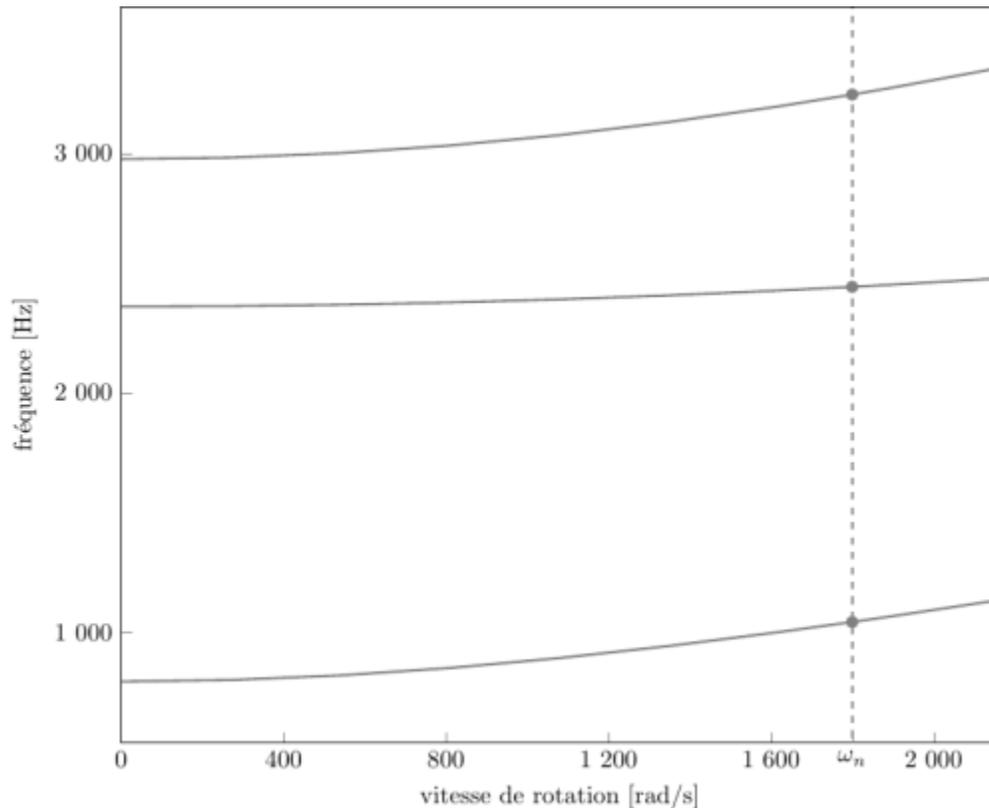
diagramme de Campbell calculé avec une précharge centrifuge linéaire, noeuds du pied d'aube encastrés (vitesse nominale ω_n = 1 800 rad/s)

- graphique (.pdf)
- données du Campbell (.csv)

Aube initiale

L'**aube initiale** est définie par des paramètres spécifiques au LAVA[5] obtenus à partir du modèle CAO de l'aube de référence. L'aube initiale est classiquement utilisée comme point de départ dans le cadre de procédures d'optimisation; sa géométrie est similaire à celle de l'aube de référence.

Fréquences propres

Fréquences des trois premiers modes (noeuds du pied d'aube encastrés),

- pour le maillage complet :

Mode	Type	Pulsation propre (rad/sec)	Fréquence propre (Hz)
1	1F	1019,1	162,2
2	1T	3539,3	563,3
3	2F	6074,0	966,7

- pour le modèle réduit :

Mode	Type	Pulsation propre (rad/sec)	Fréquence propre (Hz)
1	1F	1019,1	162,2

Mode	Type	Pulsation propre (rad/sec)	Fréquence propre (Hz)
2	1T	3540,8	563,5
3	2F	6075,8	967,0

Diagramme de Campbell

Comparaison de l'évolution des fréquences propres des 3 premiers modes, en fonction de la vitesse de rotation, pour l'aube initiale et de référence:
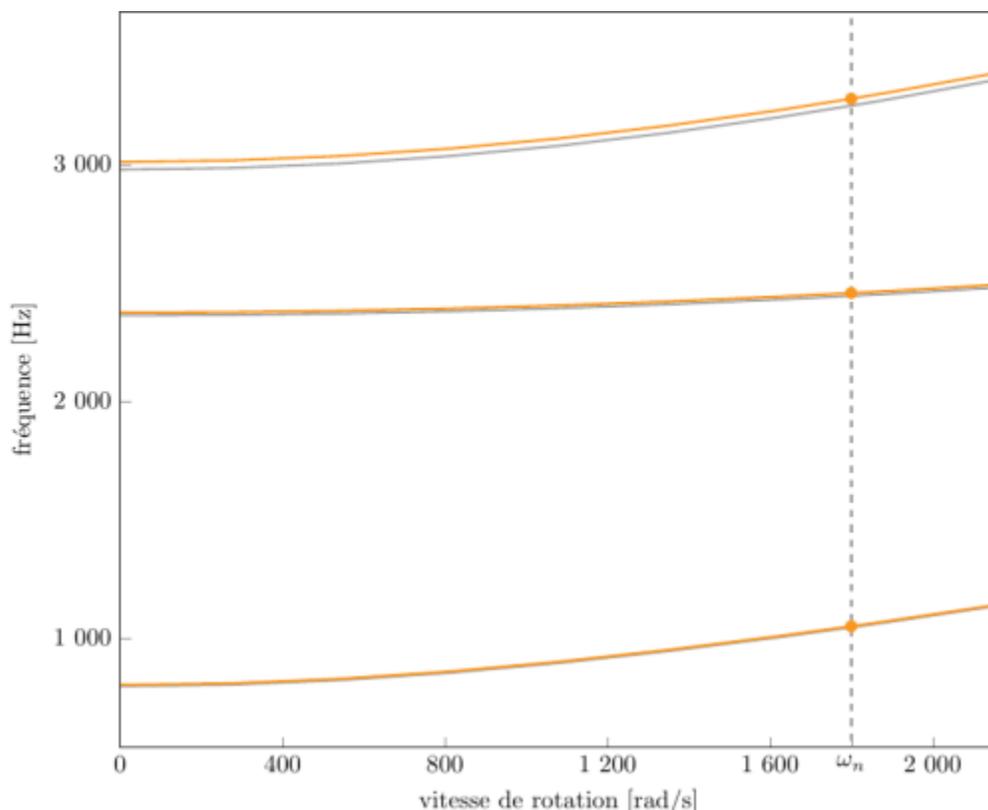
diagramme de Campbell calculé avec une précharge centrifuge linéaire, noeuds du pied d'aube encastrés (vitesse nominale ω_n = 1 800 rad/s),

aube initiale (orange), aube de référence (grise)

- graphique (.pdf)
- données du Campbell (.csv)

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1. [a, b] Reid L., Moore R. D. «Performance of Single-Stage Axial-Flow Transonic Compressor With Rotor and Stator Aspect Ratios of 1.19 and 1.26, Respectively, and With Design Pressure Ratio of 1.82 » 1978. pdf
2. [a, b, c, d, e, f, g, h, i, j] Kojtych S., Batailly A. «OpenMCAD, an open blade generator: from Multiple-Circular-Arc profiles to Computer-Aided Design model» 2022. open source code
3. [a, b] Crouse et al. «A computer program for composing compressor blading from simulated circular-arc elements on conical surfaces » 1969. NASA-TN-D-5437. pdf
4. [a, b] Reid et all. «Design and overall performance of four highly loaded, high-speed inlet stages for and

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advanced high-pressure-ratio core compressor» 1978. pdf

5. [a, b] Kojtych S. *et al*. «Methodology for the Redesign of Compressor Blades Undergoing Nonlinear Structural Interactions: Application to Blade-Tip/Casing Contacts » 2022. Journal of Engineering for Gas Turbines and Power, Vol. 145, No. 5. pdf

Document issu de la page wiki:
https://wiki.lava.polymtl.ca/public/modeles/rotor_35/accueil?rev=1722352430

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