Rotor 35

- Français
- English

Downloadable files
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Open access

Git project

Original model

Rotor 35 is part of a research program to study a advanced-core compressor design with a high compression ratio (20:1). It is therefore the first stage rotor of this eight stage transonic compressor. Of these eight stages, the first four have been designed and tested : rotors 35, 36, 37 and 38. For more information, here is a link to a report from NASA.

- Original technical report [(cite:reid1978performance>Reid L., Moore R. D. «Performance of Single-Stage Axial-Flow Transonic Compressor With Rotor and Stator Aspect Ratios of 1.19 and 1.26, Respectively, and With Design Pressure Ratio of 1.82 » 1978. pdf)]:

```
@TechReport{reid1982design,
  author           = {Reid, L. and Moore, R. D.},
  date             = {1982},
  institution      = {NASA Lewis Research Center Cleveland, OH, United
States},
  title            = {Performance of single-stage axial-flow transonic
compressor with rotor and stator aspect ratios of 1.63 and 1.78, respectively,
and with design pressure ratio of 1.82},
  number           = {NASA-TP-1974},
  url              = {https://ntrs.nasa.gov/citations/19820011348},
}
```

- Picture :



Fig. 1 https://catalog.archives.gov/id/17466807

- ```
  @Misc{brown1977records,
  author   = {Brown, M.},
  title    = {Rotor 35 - Rotor 35 - Stator 35 in casing. {R}ecords of the
  {N}ational {A}eronautics and {S}pace {A}dministration, 1903 - 2006.
  {P}hotographs relating to agency activities, facilities and personnel,
  1977 - 2013},
  url     = {https://catalog.archives.gov/id/17466807}}
  ```

Useful documents

- downloadable models (Git project)
- PDF of the NASA report :

  rotor35.pdf
- CSV file of the blade geometry :

  rotor35_original.csv

  , usable as input of OpenMCAD[(cite:Kojtych_2022_gene_aubes>Kojtych S., Batailly A. «OpenMCAD, an open blade generator: from Multiple-Circular-Arc profiles to Computer-Aided Design model» 2022. open source code)] to generate reference blade models.
  ```

Reference blade

The **reference blade** is defined with multiple-circular arc profiles[(cite:Crouse_1969_computer)] given in the original NASA report[(cite:reid1978performance)]. Corresponding models are computed with the open-source code OpenMCAD[(cite:Kojtych_2022_gene_aubes)].

Geometry

The geometry of rotor 35 is described in the original NASA report by the following tables. The length are in centimeters and the angles in degrees.

(a) For rotor 35

RP	PERCENT SPAN	RADII RI	RADII RO	BLADE ANGLES KIC	BLADE ANGLES KTC	BLADE ANGLES KOC	DELTA INC	CONE ANGLE
TIP	0.	25.248	24.511	62.55	62.99	53.21	2.09	-15.764
1	5.	24.916	24.221	61.52	61.84	52.53	2.41	-14.327
2	10.	24.571	23.931	60.55	60.74	51.87	2.72	-12.780
3	15.	24.224	23.642	59.80	59.85	51.23	2.96	-11.326
4	30.	23.163	22.772	58.34	57.74	48.54	3.41	-7.137
5	50.	21.726	21.613	56.16	54.31	44.26	4.21	-1.890
6	70.	20.221	20.454	53.70	49.53	39.16	5.51	3.545
7	85.	19.019	19.584	52.28	47.30	33.31	6.56	8.150
8	90.	18.596	19.294	52.00	46.85	30.96	6.86	9.887
9	95.	18.158	19.005	51.82	46.50	28.36	7.18	11.763
HUB	100.	17.780	18.715	51.69	46.24	25.70	7.46	12.787

RP	BLADE THICKNESSES TI	BLADE THICKNESSES TM	BLADE THICKNESSES TO	AXIAL DIMENSIONS ZI	AXIAL DIMENSIONS ZMC	AXIAL DIMENSIONS ZTC	AXIAL DIMENSIONS ZO
TIP	.025	.175	.025	.698	2.410	2.379	3.308
1	.027	.187	.027	.635	2.361	2.345	3.354
2	.028	.199	.028	.576	2.313	2.301	3.398
3	.029	.212	.029	.529	2.269	2.242	3.438
4	.032	.252	.032	.417	2.188	2.051	3.542
5	.037	.305	.038	.280	2.133	1.896	3.701
6	.042	.361	.043	.129	2.045	1.749	3.884
7	.047	.408	.047	.058	1.992	1.715	4.007
8	.048	.425	.049	.037	1.967	1.646	4.046
9	.050	.443	.050	.017	1.940	1.579	4.082
HUB	.051	.458	.051	.000	1.915	1.520	4.118

Aerodynamic design

	unit	values
pressure ratio	[-]	1.82
mass flow	[kg/s]	20.2
tip speed	[m/s]	455
tip solidity	[-]	1.3
aspect ratio	[-]	1.19
number of blades	[-]	36
rotative speed	[rad/s]	1800

Material properties

Rotor 35 is made of a 200-grade maraging steel[(cite:reid1978designandoverall>Reid *et all*. «Design and overall performance of four highly loaded, high-speed inlet stages for and advanced high-pressure-ratio core compressor» 1978. pdf)], but the exact material properties are not provided in the NASA report. The following properties are considered:

Considered properties of 18-Ni-200-maraging steel :

	unité	valeurs
alloy	[-]	18-Ni-200-maraging
Young's modulus	[GPa]	180
density	[kg/m3]	8000
Poisson's ratio	[-]	0.3
yield stress	[GPa]	1.38

CAD model

The CAD model is computed with the open source code OpenMCAD[(cite:Kojtych_2022_gene_aubes)].



pressure side



suction side

Natural frequencies

First three natural frequencies (with clamped root) for the mesh computed with OpenMCAD[(cite:Kojtych_2022_gene_aubes)]:

Mode	Type	Natural angular frequency (rad/sec)	Natural frequency (Hz)
1	1B	5000.55	795.863
2	1T	14833.91	2360.89
3	2B	18699.83	2976.17

Initial blade

The **initial blade** is defined with in-house LAVA parameters[(cite:Kojtych_2022_Methodology)] computed from the reference blade CAD model. The initial blade is usually used as starting point for an optimization process. Its geometry is similar to the one of the reference blade.

Natural frequencies

First three natural frequencies (with clamped root)

- from the whole mesh:

Mode	Type	Natural angular frequency (rad/sec)	Natural frequency (Hz)
1	1B	5037.31	801.712
2	1T	14910.06	2373.01
3	2B	18900.76	3008.15

- from the reduced order model:

Mode	Type	Natural angular frequency (rad/sec)	Natural frequency (Hz)
1	1B	5037.84	801.797
2	1T	14917.41	2374.18
3	2B	18926.96	3012.32

Fichiers téléchargeables
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Libre accès

lien vers le projet Git

Modèle original

Le rotor 35 appartient à un programme de recherche visant à étudier une conception de compresseur possédant un grand taux de compression (20:1). Il est donc le rotor du premier étage de ce compresseur transsonique de huit étages. Parmi ces huit étages, les quatre premiers ont été conçus et testés, ils correspondent aux rotors 35, 36, 37 et 38. Pour plus d'information, voici un lien vers un rapport de la NASA.

- Rapport technique original [(cite:reid1978design>)]:

```
@TechReport{reid1982design,
  author           = {Reid, L. and Moore, R. D.},
  date             = {1982},
```

```
  institution      = {NASA Lewis Research Center Cleveland, OH, United
States},
  title            = {Performance of single-stage axial-flow transonic
compressor with rotor and stator aspect ratios of 1.63 and 1.78,
respectively, and with design pressure ratio of 1.82},
  number           = {NASA-TP-1974},
  url              = {https://ntrs.nasa.gov/citations/19820011348},
}
```

- Photographie :



Fig. 1 https://catalog.archives.gov/id/17466807

- ```
  @Misc{brown1977records,
  author   = {Brown, M.},
  title    = {Rotor 35 - Rotor 35 - Stator 35 in casing. {R}ecords of the
  {N}ational {A}eronautics and {S}pace {A}dministration, 1903 - 2006.
  {P}hotographs relating to agency activities, facilities and personnel,
  1977 - 2013},
  url     = {https://catalog.archives.gov/id/17466807}}
  ```

Documents utiles

- modèles téléchargeables (lien vers projet Git)
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- PDF du rapport de la NASA :

  rotor35.pdf

- Fichier CSV de la géométrie :

  rotor35_original.csv

  , utilisable en entrée de OpenMCAD[(cite:Kojtych_2022_gene_aubes>)] pour générer l'aube de référence

Aube de référence

L'**aube de référence** est définie par des profils de type arcs circulaires multiples[(cite:Crouse_1969_computer>Crouse *et al*. «A computer program for composing compressor blading from simulated circular-arc elements on conical surfaces » 1969. NASA-TN-D-5437. pdf)], donnés dans le rapport technique original de la NASA[(cite:reid1978design)]. Les modèles associés sont obtenus avec le code en libre accès OpenMCAD[(cite:Kojtych_2022_gene_aubes)].

Géométrie

La géométrie du rotor 35 est décrite dans le rapport d'origine de la NASA par les tableaux suivants. Les grandeurs sont en centimètres et en degrés.

(a) For rotor 35

	PERCENT	RADII		BLADE ANGLES			DELTA	CONE
RP	SPAN	RI	RO	KIC	KTC	KOC	INC	ANGLE
TIP	0.	25.248	24.511	62.55	62.99	53.21	2.09	-15.764
1	5.	24.916	24.221	61.52	61.84	52.53	2.41	-14.327
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HUB	100.	17.780	18.715	51.69	46.24	25.70	7.46	12.787

	BLADE THICKNESSES			AXIAL DIMENSIONS			
RP	TI	TM	TO	ZI	ZMC	ZTC	ZO
TIP	.025	.175	.025	.698	2.410	2.379	3.308
1	.027	.187	.027	.635	2.361	2.345	3.354
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9	.050	.443	.050	.017	1.940	1.579	4.082
HUB	.051	.458	.051	.000	1.915	1.520	4.118

Caractéristiques aérodynamiques

	unités	valeurs
taux de compression	[-]	1,82
débit massique	[kg/s]	20,2
vitesse en tête	[m/s]	455
solidité en tête	[-]	1,3

	unités	valeurs
allongement	[-]	1,19
nombre d'aubes	[-]	36
vitesse de rotation	[rad/s]	1800

Propriétés matériau

Le matériau du rotor 35 est un alliage à base de nickel : un acier maraging de grade 200[(cite:reid1978designandoverall>)], mais ses caractéristiques ne sont pas fournies dans le raport de la NASA. Les propriétés considérées sont :

	unité	valeurs
alliage	[-]	18-Ni-200-maraging
module d'Young	[GPa]	180
masse volumique	[kg/m3]	8000
coefficient de Poisson	[-]	0,3
limite élastique	[GPa]	1,38

Modèle CAO

Le modèle CAO est obtenu avec OpenMCAD[(cite:Kojtych_2022_gene_aubes)].



intrados



extrados

Fréquences propres

Fréquences des trois premiers modes (noeuds du pied d'aube encastrés) pour le maillage obtenu avec OpenMCAD[(cite:Kojtych_2022_gene_aubes)] :

Mode	Type	Pulsation propre (rad/sec)	Fréquence propre (Hz)
1	1F	1018,5	162,1
2	1T	3542,5	563,8
3	2F	6094,7	970,0

Aube initiale

L'**aube initiale** est définie par des paramètres spécifiques au LAVA[(cite:Kojtych_2022_Methodology>Kojtych S. *et al*. «Methodology for the Redesign of Compressor Blades Undergoing Nonlinear Structural Interactions: Application to Blade-Tip/Casing Contacts » 2022. Journal of Engineering for Gas Turbines and Power, Vol. 145, No. 5. pdf)] obtenus à partir du modèle CAO de l'aube de référence. L'aube initiale est classiquement utilisée comme point de départ dans le cadre de procédures d'optimisation; sa géométrie est similaire à celle de l'aube de référence.

Fréquences propres

Fréquences des trois premiers modes (noeuds du pied d'aube encastrés),

- pour le maillage complet :

Mode	Type	Pulsation propre (rad/sec)	Fréquence propre (Hz)
1	1F	1019,1	162,2
2	1T	3539,3	563,3
3	2F	6074,0	966,7

- pour le modèle réduit :

Mode	Type	Pulsation propre (rad/sec)	Fréquence propre (Hz)
1	1F	1019,1	162,2
2	1T	3540,8	563,5
3	2F	6075,8	967,0

Document issu de la page wiki:
https://wiki.lava.polymtl.ca/public/modeles/rotor_35/accueil?rev=1681579686

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