Rotor 27A

- Français
- English

Downloadable files
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Open access

Git project

About

Rotor 27A is part of a research program to study the effects of aspect ratio, diffusion factor, and solidity on rotors. To do so, experimental studies have been conducted on a series of high-hub-tip-radius-ratio compressor stages representative of the middle and latter stages of axial-flow compressors. In fact, 14 middle stages were tested to assess the effects on performance of varying both diffusion through the rotor and stator blades and blade aspect ratio. Among these 14 stages, there are rotors 23B, 24A, 25A, 26B, 27A and 28B. Both the tip diameter and the hub-tip radius ratio were held constant throughout each stage at 50.8 centimeters and 0.8, respectively.

- Original technical report [(cite:britsch1979design>Britsch. «Design and overall performance of four highly loaded, high speed inlet stages for an advanced high-pressure-ratio core compressor » 1979. pdf)]:

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@TechReport{britsch1979design,
  author           = {Britsch, Werner R. and Osborn, Walter M. and
Laessig, Mark R.},
  date             = {1979},
  institution      = {NASA Lewis Research Center Cleveland, OH, United
States},
  title            = {Effects of Diffusion Factor, Aspect Ratio, and
Solidity on Overall Performance of 14 Compressor Middle Stages},
  number           = {NASA-TP-1523},
  url              = {https://ntrs.nasa.gov/citations/19790025039},
}
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- Picture :



https://catalog.archives.gov/id/17447129

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@Misc{brown1975records,
author   = {Brown, M.},
title    = {Rotor 27. {R}ecords of the {N}ational {A}eronautics and {S}pace
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facilities and personnel, 1975 - 2013},
note     =
{\href{https://catalog.archives.gov/id/17447129}{https://catalog.archives.gov/
id/17447129}, 1975 }, % for Fig. 1}
```

Useful documents

- downloadable models (Git project)
  - NASA technical report (.pdf)
  - geometrical parameters file (.csv), usable as input of OpenMCAD[(cite:Kojtych_2022_gene_aubes>Kojtych S., Batailly A. «OpenMCAD, an open blade generator: from Multiple-Circular-Arc profiles to Computer-Aided Design model» 2022. code en libre accès)] to generate reference blade models.

Reference blade

The **reference blade** is defined with multiple-circular arc profiles[(cite:Crouse_1969_computer)] given in

the original NASA report[(cite:britsch1979design)]. Corresponding models are computed with the open-source code OpenMCAD[(cite:Kojtych_2022_gene_aubes)].

Geometry

The geometry of rotor 27A is described in the original NASA report by the following tables. The length are in centimeters and the angles in degrees.

TABLE 12. – BLADE GEOMETRY FOR ROTOR 27A

RP	PERCENT SPAN	RADII RI	RADII RO	BLADE ANGLES KIC	BLADE ANGLES KTC	BLADE ANGLES KOC	DELTA INC	CONE ANGLE
TIP	0.	25.400	25.400	63.52	56.76	49.69	2.61	0.057
1	5.	25.177	25.146	63.19	56.32	48.85	2.73	-0.446
2	10.	24.927	24.892	62.85	55.81	47.95	2.96	-0.501
3	15.	24.673	24.638	62.52	55.28	46.95	3.15	-0.491
4	30.	23.895	23.876	61.64	53.78	43.96	3.72	-0.259
5	50.	22.852	22.860	60.54	51.80	39.60	4.45	0.107
6	70.	21.806	21.844	59.40	49.68	34.47	5.13	0.451
7	85.	21.027	21.082	58.57	47.95	29.81	5.60	0.621
8	90.	20.773	20.828	58.29	47.25	27.79	5.74	0.606
9	95.	20.525	20.574	58.02	46.39	25.19	5.87	0.525
HUB	100.	20.320	20.320	57.79	45.61	22.37	5.98	0.057

RP	BLADE THICKNESSES TI	BLADE THICKNESSES TM	BLADE THICKNESSES TO	AXIAL DIMENSIONS ZIC	AXIAL DIMENSIONS ZMC	AXIAL DIMENSIONS ZTC	AXIAL DIMENSIONS ZOC
TIP	0.051	0.213	0.051	0.609	2.388	2.330	4.524
1	0.051	0.224	0.051	0.586	2.388	2.305	4.560
2	0.051	0.236	0.051	0.559	2.388	2.276	4.598
3	0.051	0.248	0.051	0.532	2.387	2.245	4.638
4	0.051	0.284	0.051	0.453	2.384	2.148	4.754
5	0.051	0.330	0.051	0.348	2.381	2.009	4.914
6	0.051	0.375	0.051	0.229	2.374	1.849	5.086
7	0.051	0.406	0.051	0.131	2.368	1.716	5.231
8	0.051	0.416	0.051	0.092	2.366	1.665	5.289
9	0.051	0.426	0.051	0.044	2.362	1.606	5.361
HUB	0.051	0.434	0.051	-0.000	2.359	1.554	5.438

Aerodynamic design

	unit	value
pressure ratio	[-]	1.328
mass flow	[kg/s]	9.46
tip speed	[m/s]	243.8
tip solidity	[-]	1.8
aspect ratio	[-]	0.7
number of blades	[-]	40
rotative speed	[rad/s]	960.28

Material properties

The original material of the rotor 27A is not defined in the NASA report.

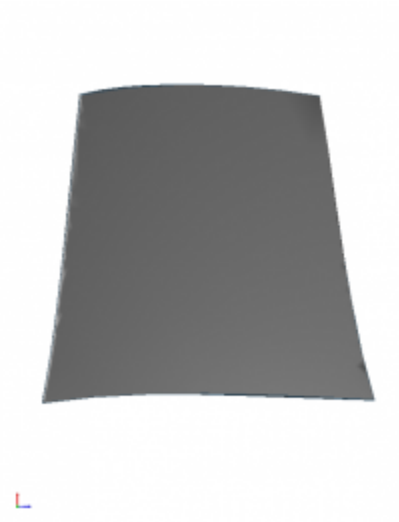
Considered properties: 200-grade maraging steel :

	unit	value

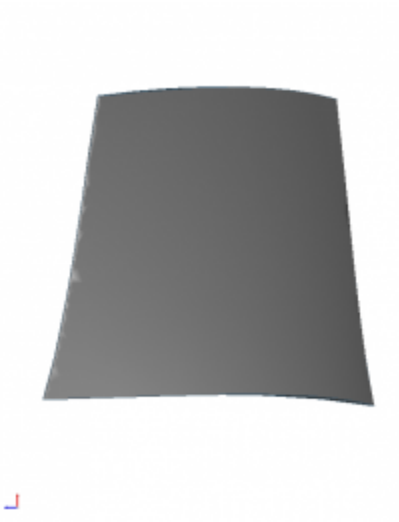
	unit	value
alloy	[-]	18-Ni-200-maraging
Young's modulus	[GPa]	180
density	[kg/m3]	8000
Poisson's ratio	[-]	0.3
yield stress	[GPa]	1.38

CAD model

The CAD model is computed with the open source code OpenMCAD[(cite:Kojtych_2022_gene_aubes)].



pressure side



suction side

Natural frequencies

First three natural frequencies (with clamped root) for the mesh computed with OpenMCAD[(cite:Kojtych_2022_gene_aubes)]:

Mode	Type	Natural angular frequency (rad/sec)	Natural frequency (Hz)
1	1B	8735.07	1390.23
2	1T	17430.88	2774.21
3	2B	27068.03	4308.01

Initial blade

The **initial blade** is defined with in-house LAVA parameters[(cite:Kojtych_2022_Methodology)] computed from the reference blade CAD model. The initial blade is usually used as starting point for an optimization process. Its geometry is similar to the one of the reference blade.

Natural frequencies

First three natural frequencies (with clamped root)

- from the whole mesh:

Mode	Type	Natural angular frequency (rad/sec)	Natural frequency (Hz)
1	1B	8708.18	1385.95
2	1T	17470.52	2780.52
3	2B	27124.26	4316.96

- from the reduced order model:

Mode	Type	Natural angular frequency (rad/sec)	Natural frequency (Hz)
1	1B	8710.32	1386.29
2	1T	17487.68	2783.25
3	2B	27168.93	4324.07

Fichiers téléchargeables
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Libre accès

lien vers le projet Git

À propos

Le rotor 27A fait partie d'un programme de recherche visant à étudier les effets de l’allongement, du facteur de diffusion et de la solidité des rotors. Pour ce faire, des études expérimentales ont été menées sur une série d'étages de compresseurs à fort rapport entre les rayons du moyeu et de la tête d’aube, représentatifs des étages moyens et avancés des compresseurs à flux axial. En effet, 14 étages intermédiaires ont été testés pour évaluer les effets sur les performances de la variation de la diffusion et de l’allongement des aubes. Parmi ces 14 étages, on trouve les rotors 23B, 24A, 25A, 26B, 27A et 28B. Le diamètre de l'extrémité des aubes et le rapport entre les rayons du moyeu et de la tête d’aube ont été maintenus constants tout au long de chaque étage à 50.8 centimètres et 0.8, respectivement.

- Rapport technique original [(cite:britsch1979design>Britsch. «Design and overall performance of four highly loaded, high speed inlet stages for an advanced high-pressure-ratio core compressor »

1979. pdf)]:

```
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- Photographie :



https://catalog.archives.gov/id/17447129

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note     =
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id/17447129}, 1975 }, % for Fig. 1}
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Documents utiles

- modèles téléchargeables (lien vers projet Git)
  - rapport technique original de la NASA (.pdf)
  - fichier de paramètres géométriques (.csv), utilisable en entrée de OpenMCAD[(cite:Kojtych_2022_gene_aubes>Kojtych S., Batailly A. «OpenMCAD, an open blade generator: from Multiple-Circular-Arc profiles to Computer-Aided Design model» 2022. code en libre accès)] pour générer l'aube de référence

Aube de référence

L'**aube de référence** est définie par des profils de type arcs circulaires multiples[(cite:Crouse_1969_computer>Crouse *et al*. «A computer program for composing compressor blading from simulated circular-arc elements on conical surfaces » 1969. NASA-TN-D-5437. pdf)], donnés dans le rapport technique original de la NASA[(cite:britsch1979design)]. Les modèles associés sont obtenus avec le code en libre accès OpenMCAD[(cite:Kojtych_2022_gene_aubes)].

Géométrie

Le matériau original du rotor 27A n'est pas défini dans le rapport de la NASA.

Propriétés considérées : un acier maraging de grade 200 :

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6	70.	21.806	21.844	59.40	49.68	34.47	5.13	0.451
7	85.	21.027	21.082	58.57	47.95	29.81	5.60	0.621
8	90.	20.773	20.828	58.29	47.25	27.79	5.74	0.606
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6	0.051	0.375	0.051	0.229	2.374	1.849	5.086
7	0.051	0.406	0.051	0.131	2.368	1.716	5.231
8	0.051	0.416	0.051	0.092	2.366	1.665	5.289
9	0.051	0.426	0.051	0.044	2.362	1.606	5.361
HUB	0.051	0.434	0.051	-0.000	2.359	1.554	5.438

Caractéristiques aérodynamiques

	unités	valeurs
taux de compression	[-]	1,328
débit massique	[kg/s]	9,46
vitesse en tête	[m/s]	243,8
solidité en tête	[-]	1,8
allongement	[-]	0,7
nombre d'aubes	[-]	40
vitesse de rotation	[rad/s]	960,28

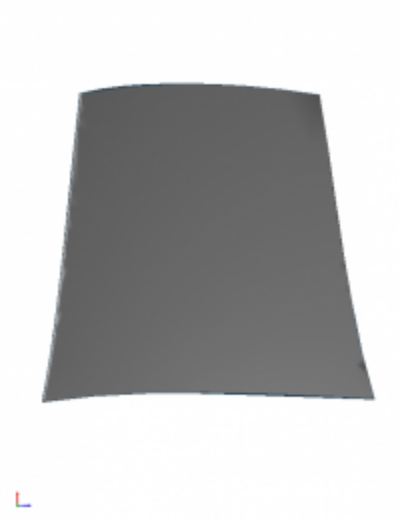
Propriétés matériau

Le matériau supposé du rotor 23B est un alliage à base de nickel : un acier maraging de grade 200

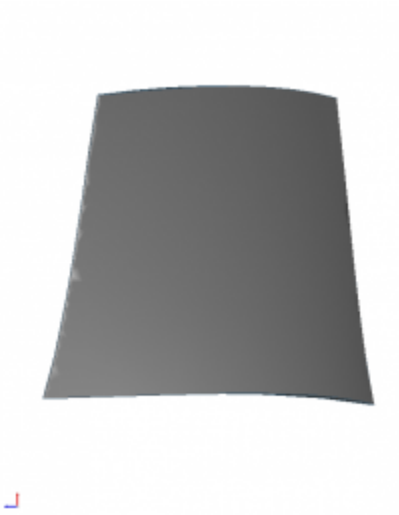
	unité	valeurs
alliage	[-]	18-Ni-200-maraging
module d'Young	[GPa]	180
masse volumique	[kg/m3]	8000
coefficient de Poisson	[-]	0,3
limite élastique	[GPa]	1,38

Modèle CAO

Le modèle CAO est obtenu avec OpenMCAD[(cite:Kojtych_2022_gene_aubes)].



intrados



extrados

Fréquences propres

Fréquences des trois premiers modes (noeuds du pied d'aube encastrés) pour le maillage obtenu avec OpenMCAD[(cite:Kojtych_2022_gene_aubes)] :

Mode	Type	Pulsation propre (rad/sec)	Fréquence propre (Hz)
1	1F	8735,07	1390,23
2	1T	17430,88	2774,21
3	2F	27068,03	4308,01

Aube initiale

L'**aube initiale** est définie par des paramètres spécifiques au LAVA[(cite:Kojtych_2022_Methodology>Kojtych S. *et al*. «Methodology for the Redesign of Compressor Blades Undergoing Nonlinear Structural Interactions: Application to Blade-Tip/Casing Contacts » 2022. Journal of Engineering for Gas Turbines and Power, Vol. 145, No. 5. pdf)] obtenus à partir du modèle CAO de l'aube de référence. L'aube initiale est classiquement utilisée comme point de départ dans le cadre de

procédures d'optimisation; sa géométrie est similaire à celle de l'aube de référence.

Fréquences propres

Fréquences des trois premiers modes (noeuds du pied d'aube encastrés),

- pour le maillage complet :

Mode	Type	Pulsation propre (rad/sec)	Fréquence propre (Hz)
1	1F	8708,18	1385,95
2	1T	17470,52	2780,52
3	2F	27124,26	4316,96

- pour le modèle réduit :

Mode	Type	Pulsation propre (rad/sec)	Fréquence propre (Hz)
1	1F	8710,32	1386,29
2	1T	17487,68	2783,25
3	2F	27168,93	4324,07

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Document issu de la page wiki:
https://wiki.lava.polymtl.ca/public/modeles/rotor_27a/accueil?rev=1679275944

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