Rotor 26B

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Original model

Rotor 26B is part of a research program to study the effects of aspect ratio, diffusion factor, and solidity on rotors. To do so, experimental studies have been conducted on a series of high-hub-tip-radius-ratio compressor stages representative of the middle and latter stages of axial-flow compressors. In fact, 14 middle stages were tested to assess the effects on performance of varying both diffusion through the rotor and stator blades and blade aspect ratio. Among these 14 stages, there are rotors 23B, 24A, 25A, 26B, 27A and 28B. Both the tip diameter and the hub-tip radius ratio were held constant throughout each stage at 50.8 centimeters and 0.8, respectively.

- Original technical report [(cite:britsch1979design>Britsch. «Design and overall performance of four highly loaded, high speed inlet stages for an advanced high-pressure-ratio core compressor » 1979. pdf)]:

```
@TechReport{britsch1979design,
author      = {Britsch, Werner R. and Osborn, Walter M. and Laessig, Mark
R.},
title       = {Effects of Diffusion Factor, Aspect Ratio, and Solidity on
Overall Performance of 14 Compressor Middle Stages},
institution = {NASA Lewis Research Center Cleveland, OH, United States},
note        = {NASA-TP-1523, url~:
\url{https://ntrs.nasa.gov/citations/19790025039}, 1979 }}
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- Picture :



https://catalog.archives.gov/id/17448811

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@Misc{huebler1975records,
author   = {Huebler, D.},
title    = {Rotor 26B stator ring half stator 21. {R}ecords of the {N}ational
{A}eronautics and {S}pace {A}dministration, 1903 - 2006. {P}hotographs
relating to agency activities, facilities and personnel, 1975 - 2013},
note     =
{\href{https://catalog.archives.gov/id/17448811}{https://catalog.archives.gov/
id/17448811}, 1975 }, % for Fig. 1}
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Useful documents

- PDF of the NASA report : rotor26b.pdf
- CSV file of the blade geometry : rotor26b_original.csv

Geometry

The geometry of rotor 26B is described in the original NASA report by the following tables. The length are in centimeters and the angles in degrees.

TABLE 11. – BLADE GEOMETRY FOR ROTOR 26B

RP	PERCENT SPAN	RADII RI	RADII RO	BLADE ANGLES KIC	BLADE ANGLES KTC	BLADE ANGLES KOC	DELTA INC.	CONE ANGLE
TIP	0.	25.400	25.400	64.09	56.86	49.41	2.65	0.057
1	5.	25.176	25.146	63.76	56.43	48.59	2.22	-0.747
2	10.	24.926	24.892	63.41	55.94	47.71	2.41	-0.836
3	15.	24.672	24.638	63.08	55.41	46.75	2.60	-0.814
4	30.	23.895	23.876	62.19	53.94	43.79	3.17	-0.427
5	50.	22.851	22.860	61.08	51.97	39.45	3.90	0.186
6	70.	21.807	21.844	59.94	49.85	34.32	4.59	0.758
7	85.	21.028	21.082	59.09	48.12	29.64	5.06	1.038
8	90.	20.774	20.828	58.82	47.42	27.61	5.21	1.017
9	95.	20.526	20.574	58.54	46.55	25.01	5.34	0.685
HUB	100.	20.320	20.320	58.31	45.77	22.18	5.45	0.057

RP	BLADE THICKNESSES TI	BLADE THICKNESSES TM	BLADE THICKNESSES TO	AXIAL DIMENSIONS ZIC	AXIAL DIMENSIONS ZMC	AXIAL DIMENSIONS ZTC	AXIAL DIMENSIONS ZOC
TIP	0.051	0.126	0.051	0.339	1.369	1.345	2.022
1	0.051	0.132	0.051	0.326	1.370	1.331	2.845
2	0.051	0.139	0.051	0.312	1.371	1.315	2.666
3	0.051	0.146	0.051	0.297	1.371	1.298	2.600
4	0.051	0.167	0.051	0.254	1.373	1.244	2.759
5	0.051	0.194	0.051	0.195	1.374	1.166	2.856
6	0.051	0.220	0.051	0.129	1.374	1.078	2.991
7	0.051	0.239	0.051	0.074	1.373	1.005	3.050
8	0.051	0.245	0.051	0.052	1.373	0.974	3.085
9	0.051	0.251	0.051	0.024	1.372	0.941	3.129
HUB	0.051	0.255	0.051	-0.000	1.372	0.911	3.175

Aerodynamic design

	unit	values
pressure ratio	[-]	1.328
mass flow	[kg/s]	9.46
tip speed	[m/s]	243.8
tip solidity	[-]	1.8
aspect ratio	[-]	1.2
number of blades	[-]	68
rotative speed	[rad/s]	960.28

Material properties

The original material of the rotor 26B is not defined in the NASA report.

Considered properties: 200-grade maraging steel :

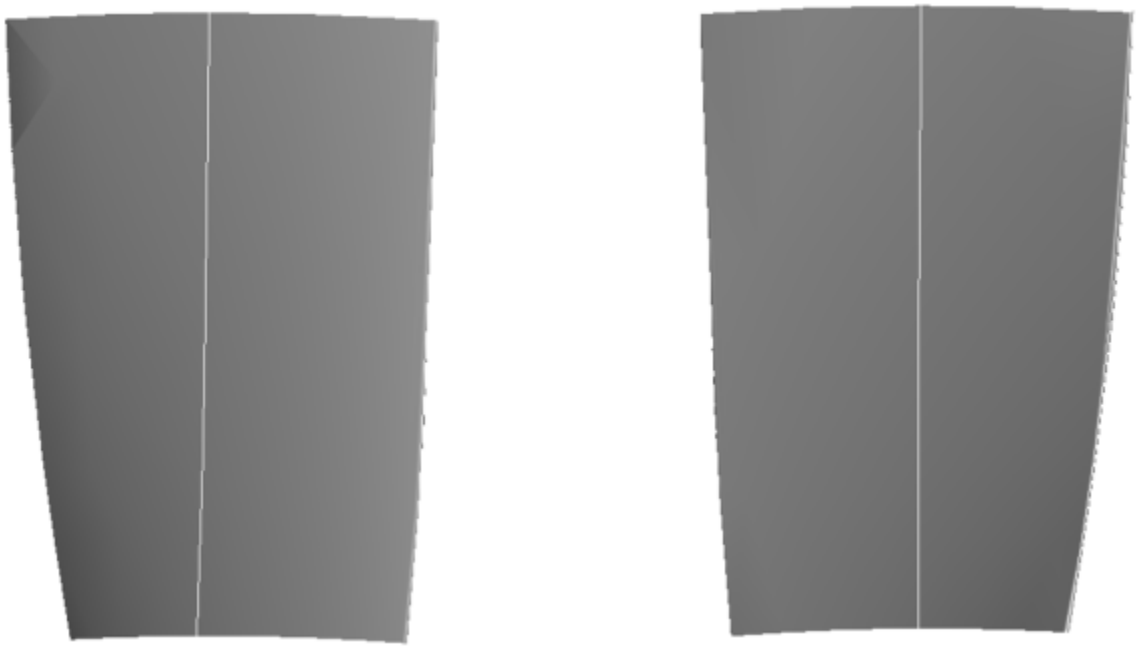
	unité	valeurs
alloy	[-]	18-Ni-200-maraging
Young's modulus	[GPa]	180
density	[kg/m3]	8000
Poisson's ratio	[-]	0.3
yield stress	[GPa]	1.38

First three natural frequencies (with clamped root) for the mesh:

1. (1B): 5670.9 rad/s / 902.5 Hz

2. (1T): 13737.9rad/s / 2186.4 Hz
3. (2B): 24172.4 rad/s / 3847.1 Hz

CAD



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Modèle original

Le rotor 26B fait partie d'un programme de recherche visant à étudier les effets de l'allongement, du facteur de diffusion et de la solidité des rotors. Pour ce faire, des études expérimentales ont été menées sur une série d'étages de compresseurs à fort rapport entre les rayons du moyeu et de la tête d'aube, représentatifs des étages moyens et avancés des compresseurs à flux axial. En effet, 14 étages intermédiaires ont été testés pour évaluer les effets sur les performances de la variation de la diffusion et de l'allongement des aubes. Parmi ces 14 étages, on trouve les rotors 23B, 24A, 25A, 26B, 27A et 28B. Le diamètre de l'extrémité des aubes et le rapport entre les rayons du moyeu et de la tête d'aube ont été maintenus constants tout au long de chaque étage à 50.8 centimètres et 0.8, respectivement.

- Rapport technique original [(cite:britsch1979design>Britsch. «Design and overall performance of four highly loaded, high speed inlet stages for an advanced high-pressure-ratio core compressor » 1979. pdf)]:

```
@TechReport{britsch1979design,
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- Photographie :



https://catalog.archives.gov/id/17448811

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note     =
{\href{https://catalog.archives.gov/id/17448811}{https://catalog.archives.gov/
id/17448811}, 1975 }, % for Fig. 1}
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Documents utiles

- PDF du rapport de la NASA :

rotor26b.pdf

- Fichier CSV de la géométrie :

rotor26b_original.csv

Géométrie

La géométrie du rotor 26B est décrite dans le rapport d'origine de la NASA par les tableaux suivants. Les grandeurs sont en centimètres et en degrés.

TABLE 11. – BLADE GEOMETRY FOR ROTOR 26B

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5	50.	22.851	22.860	61.08	51.97	39.45	3.90	0.186
6	70.	21.807	21.844	59.94	49.85	34.32	4.59	0.758
7	85.	21.028	21.082	59.09	48.12	29.64	5.08	1.038
8	90.	20.774	20.828	58.82	47.42	27.61	5.21	1.017
9	95.	20.526	20.574	58.54	46.55	25.01	5.34	0.885
HUB	100.	20.320	20.320	58.31	45.77	22.18	5.45	0.057

RP	BLADE THICKNESSES TI	BLADE THICKNESSES TM	BLADE THICKNESSES TO	AXIAL DIMENSIONS ZIC	AXIAL DIMENSIONS ZMC	AXIAL DIMENSIONS ZTC	AXIAL DIMENSIONS ZOC
TIP	0.051	0.126	0.051	0.339	1.369	1.345	2.022
1	0.051	0.132	0.051	0.326	1.370	1.331	2.645
2	0.051	0.139	0.051	0.312	1.371	1.315	2.666
3	0.051	0.146	0.051	0.297	1.371	1.298	2.600
4	0.051	0.167	0.051	0.254	1.373	1.244	2.759
5	0.051	0.194	0.051	0.195	1.374	1.166	2.838
6	0.051	0.220	0.051	0.129	1.374	1.078	2.961
7	0.051	0.239	0.051	0.074	1.373	1.005	3.050
8	0.051	0.245	0.051	0.052	1.373	0.974	3.086
9	0.051	0.251	0.051	0.024	1.372	0.941	3.129
HUB	0.051	0.255	0.051	-0.000	1.372	0.911	3.175

Caractéristiques aérodynamiques

	unités	valeurs
taux de compression	[-]	1,328
débit massique	[kg/s]	9,46
vitesse en tête	[m/s]	243,8
solidité en tête	[-]	1,8
allongement	[-]	1,2
nombre d'aubes	[-]	68
vitesse de rotation	[rad/s]	960,28

Propriétés matériau

Le matériau original du rotor 26B n'est pas défini dans le rapport de la NASA.

Propriétés considérées : un acier maraging de grade 200 :

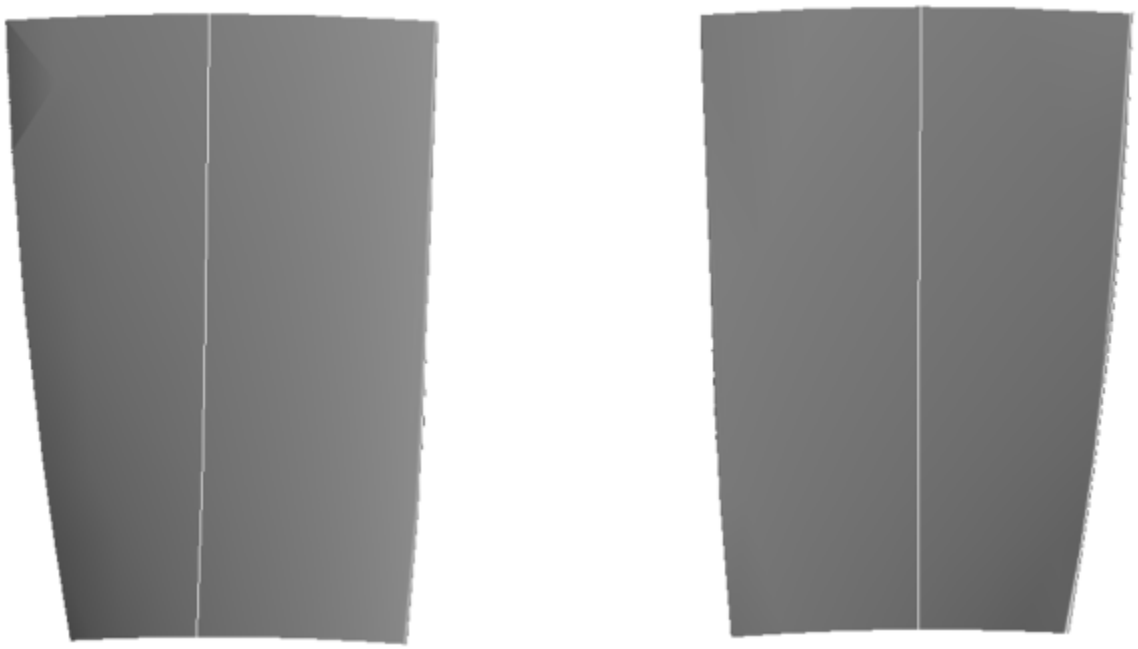
	unité	valeurs
alliage	[-]	18-Ni-200-maraging
module d'Young	[GPa]	180

	unité	valeurs
masse volumique	[kg/m3]	8000
coefficient de Poisson	[-]	0,3
limite élastique	[GPa]	1,38

Fréquences des trois premiers modes (noeuds de la base encastrés) pour le maillage :

1. (1B): 5670,9 rad/s / 902,5 Hz
2. (1T): 13737,9rad/s / 2186,4 Hz
3. (2B): 24172,4 rad/s / 3847,1 Hz

CAO



Document issu de la page wiki:
https://wiki.lava.polymtl.ca/public/modeles/rotor_26b/accueil?rev=1663351738

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