

# Rotor 18

- [Français](#)
- [English](#)

Downloadable files

×

**Open access**

[Git project](#)

## About

One way to reduce compressor weight is to obtain a high pressure ratio per stage, thereby reducing the number of stages. As a consequence, several single-stage fans have been designed to produce stage pressure ratios ranging from 1.9 to 2.2. Rotor 18 is one of those fans with a pressure ratio of 1.925.

- Original technical report <sup>[1]</sup>:

```
@TechReport{lewis1974design,
  author      = {Lewis, George W. and Reid, Lonnie and Tysl, Edward
R.},
  date        = {1974},
  institution = {NASA Lewis Research Center Cleveland, OH, United
States},
  title       = {Design and performance of a high-pressure-ratio,
highly loaded axial-flow transonic compressor stage},
  number      = {NASA-TM X-3100},
  url         = {https://ntrs.nasa.gov/citations/19740025108},
}
```

- Picture :



Fig1. <https://catalog.archives.gov/id/17422714>

### Useful documents

- [downloadable models](#) (Git project)
  - NASA technical report (.pdf)
  - geometrical parameters file (.csv), usable as input of OpenMCAD<sup>[2]</sup> to generate reference blade models.

## Reference blade

The **reference blade** is defined with multiple-circular arc profiles<sup>[3]</sup> given in the original NASA report<sup>[1]</sup>. Corresponding models are computed with the open-source code OpenMCAD<sup>[2]</sup>.

## Geometry

[The geometry of rotor 18 is described in the original NASA report](#) by the following tables. The length are in centimeters and the angles in degrees.

TABLE IV. - BLADE GEOMETRY FOR ROTOR 18

| RP  | PERCENT |        | RADI   |  | BLADE ANGLES |       |        | DELTA | CONE    |
|-----|---------|--------|--------|--|--------------|-------|--------|-------|---------|
|     | SPAN    | R1     | R0     |  | KIC          | KTC   | KOC    | [NC   | ANGLE   |
| TIP | 0.      | 25.019 | 24.359 |  | 63.50        | 60.06 | 48.60  | 2.49  | -15.536 |
| 1   | 5.      | 24.442 | 23.871 |  | 61.91        | 58.53 | 47.71  | 2.75  | -12.940 |
| 2   | 10.     | 23.906 | 23.383 |  | 60.51        | 57.10 | 46.55  | 3.00  | -11.383 |
| 3   | 30.     | 21.655 | 21.433 |  | 55.62        | 51.09 | 39.82  | 4.08  | -4.201  |
| 4   | 48.     | 19.590 | 19.726 |  | 52.08        | 45.52 | 31.18  | 5.03  | 2.263   |
| 5   | 50.     | 19.288 | 19.482 |  | 51.61        | 44.86 | 29.62  | 5.17  | 3.196   |
| 6   | 53.     | 18.983 | 19.238 |  | 51.14        | 44.24 | 27.97  | 5.30  | 4.131   |
| 7   | 55.     | 18.676 | 18.994 |  | 50.68        | 43.63 | 26.26  | 5.43  | 5.076   |
| 8   | 58.     | 18.367 | 18.750 |  | 50.22        | 42.97 | 24.48  | 5.56  | 6.033   |
| 9   | 70.     | 16.777 | 17.531 |  | 48.02        | 39.76 | 14.38  | 6.16  | 10.944  |
| 10  | 90.     | 14.079 | 15.580 |  | 44.63        | 35.58 | -7.70  | 6.86  | 19.179  |
| 11  | 95.     | 13.384 | 15.093 |  | 43.76        | 34.81 | -14.81 | 6.97  | 21.167  |
| HUB | 100.    | 12.700 | 14.605 |  | 42.89        | 34.15 | -22.54 | 7.04  | 22.937  |

| RP  | BLADE THICKNESSES |       |       | AXIAL DIMENSIONS |       |       |       |
|-----|-------------------|-------|-------|------------------|-------|-------|-------|
|     | T1                | TM    | TO    | ZIC              | ZMC   | ZTC   | ZOC   |
| TIP | 0.051             | 0.152 | 0.051 | 1.148            | 2.212 | 2.267 | 3.523 |
| 1   | 0.051             | 0.163 | 0.051 | 1.087            | 2.213 | 2.225 | 3.573 |
| 2   | 0.051             | 0.173 | 0.051 | 1.031            | 2.213 | 2.183 | 3.626 |
| 3   | 0.051             | 0.216 | 0.051 | 0.812            | 2.205 | 1.966 | 3.847 |
| 4   | 0.051             | 0.256 | 0.051 | 0.627            | 2.193 | 1.719 | 4.048 |
| 5   | 0.051             | 0.262 | 0.051 | 0.602            | 2.191 | 1.681 | 4.076 |
| 6   | 0.051             | 0.267 | 0.051 | 0.578            | 2.188 | 1.643 | 4.125 |
| 7   | 0.051             | 0.273 | 0.051 | 0.554            | 2.185 | 1.604 | 4.133 |
| 8   | 0.051             | 0.279 | 0.051 | 0.529            | 2.182 | 1.563 | 4.160 |
| 9   | 0.051             | 0.309 | 0.051 | 0.396            | 2.164 | 1.341 | 4.296 |
| 10  | 0.051             | 0.360 | 0.051 | 0.149            | 2.126 | 0.924 | 4.466 |
| 11  | 0.051             | 0.373 | 0.051 | 0.076            | 2.117 | 0.807 | 4.488 |
| HUB | 0.051             | 0.386 | 0.051 | 0.000            | 2.107 | 0.687 | 4.532 |

## Aerodynamic design

|                                   | unit    | value |
|-----------------------------------|---------|-------|
| pressure ratio                    | [-]     | 1.925 |
| mass flow                         | [kg/s]  | 29.5  |
| tip speed                         | [m/s]   | 422   |
| tip solidity                      | [-]     | 1.7   |
| aspect ratio                      | [-]     | 2.6   |
| number of blades                  | [-]     | 56    |
| nominal rotation speed $\omega_n$ | [rad/s] | 1686  |

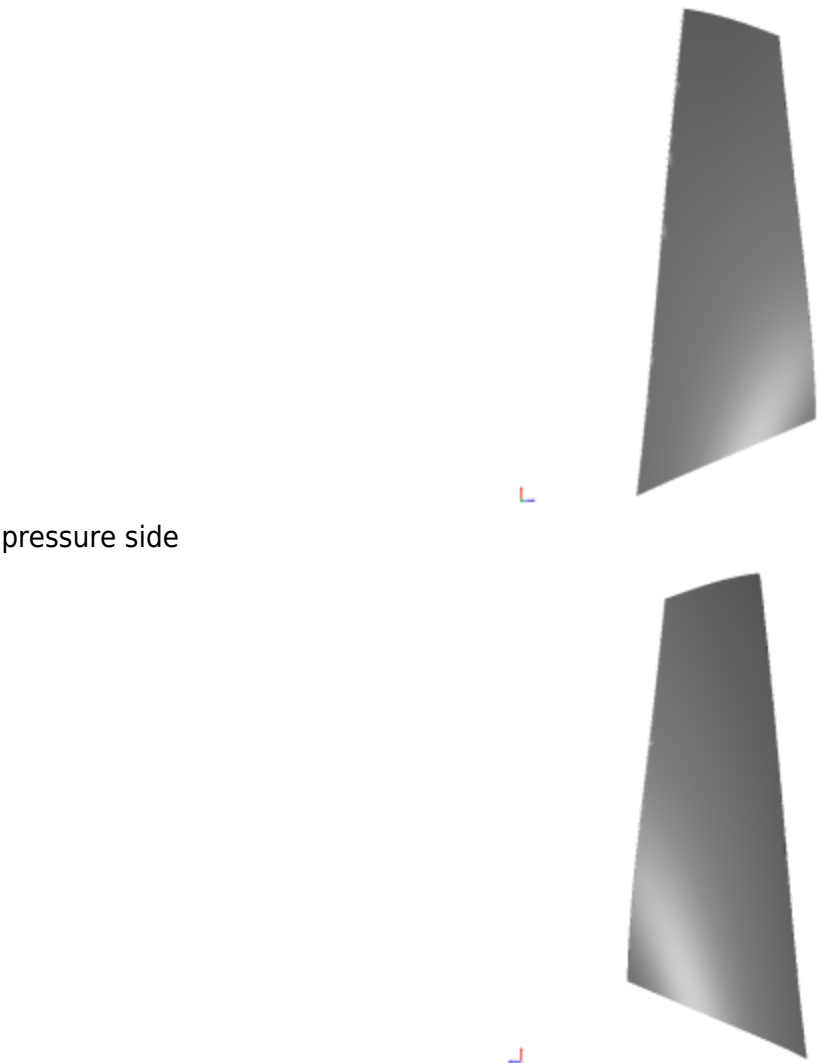
## Material properties

The original material of the rotor 18 is not defined in the NASA report. A generic titanium Ti-6Al-4V is considered:

|                 | unit                 | value     |
|-----------------|----------------------|-----------|
| alloy           | [-]                  | Ti-6Al-4V |
| Young's modulus | [GPa]                | 108       |
| density         | [kg/m <sup>3</sup> ] | 4400      |
| Poisson's ratio | [-]                  | 0.34      |
| yield stress    | [GPa]                | 0.824     |

## CAD model

The CAD model is computed with the open source code OpenMCAD<sup>[2]</sup>.



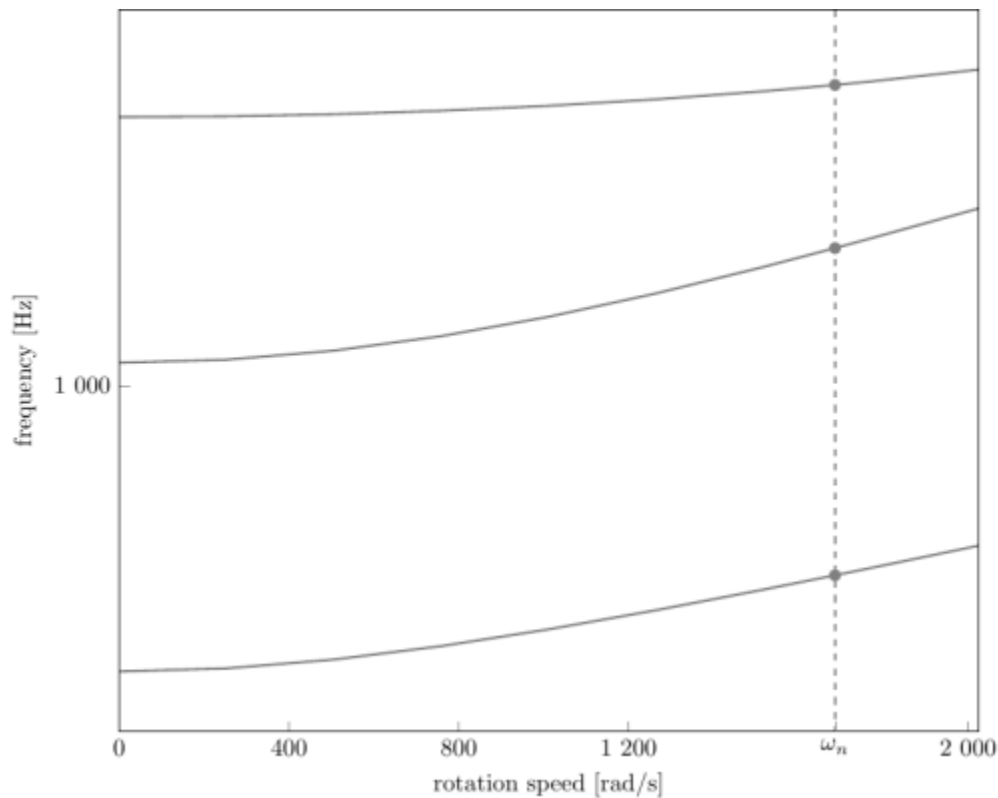
## Natural frequencies

First three natural frequencies (with clamped root) for the mesh computed with OpenMCAD<sup>[2]</sup>:

| Mode | Type | Natural angular frequency (rad/sec) | Natural frequency (Hz) |
|------|------|-------------------------------------|------------------------|
| 1    | 1B   | 2023.14                             | 321.99                 |
| 2    | 2B   | 6631.02                             | 1055.36                |
| 3    | 1T   | 10297.70                            | 1638.93                |

## Campbell diagram

Evolution of the natural frequencies of the first 3 vibration modes, as a function of rotation speed, for the mesh computed with OpenMCAD<sup>[2]</sup>:



Campbell diagram computed with a linear centrifugal preload, with clamped root (nominal rotation speed  $\omega_n = 1\,686$  rad/s)

- graph (.pdf)
- Campbell data (.csv)

## Initial blade

The **initial blade** is defined with in-house LAVA parameters<sup>[4]</sup> computed from the reference blade CAD model. The initial blade is usually used as starting point for an optimization process. Its geometry is similar to the one of the reference blade.

## Natural frequencies

First three natural frequencies (with clamped root)

- from the whole mesh:

| Mode | Type | Natural angular frequency (rad/sec) | Natural frequency (Hz) |
|------|------|-------------------------------------|------------------------|
| 1    | 1B   | 2018.44                             | 321.244                |
| 2    | 2B   | 6628.45                             | 1054.95                |
| 3    | 1T   | 10299.84                            | 1639.27                |

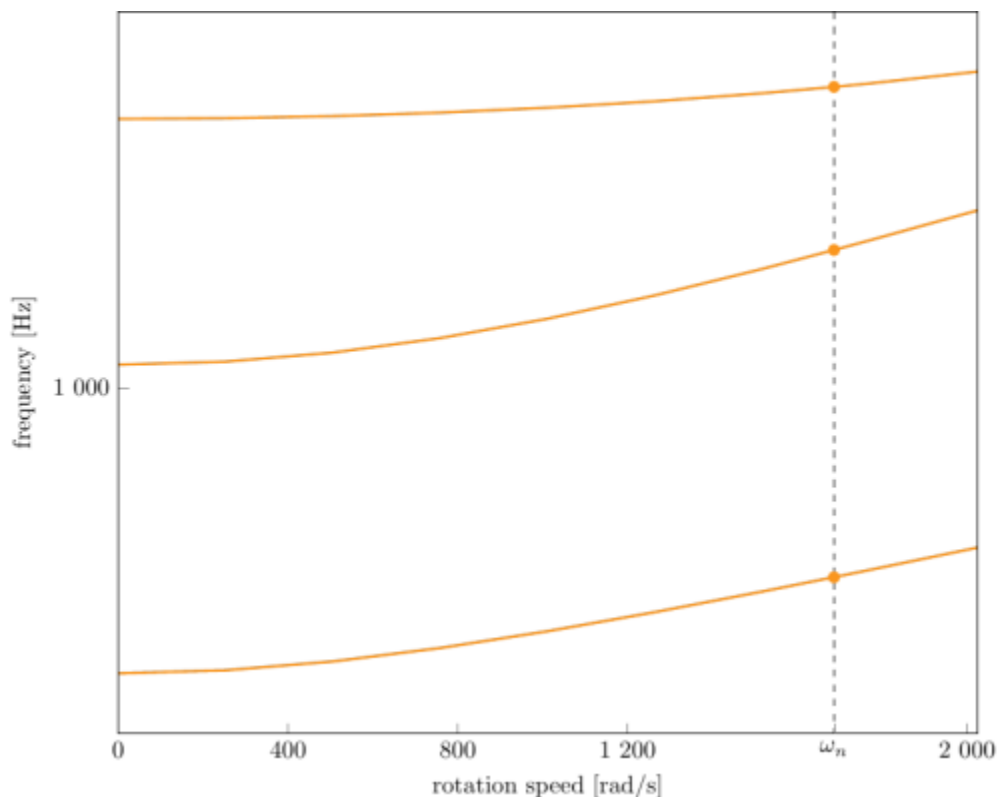
- from the reduced order model:

| Mode | Type | Natural angular frequency (rad/sec) | Natural frequency (Hz) |
|------|------|-------------------------------------|------------------------|
| 1    | 1B   | 2018.52                             | 321.258                |
| 2    | 2B   | 6631.02                             | 1055.36                |

| Mode | Type | Natural angular frequency (rad/sec) | Natural frequency (Hz) |
|------|------|-------------------------------------|------------------------|
| 3    | 1T   | 10304.68                            | 1640.04                |

## Campbell diagram

Comparison of the evolution of the natural frequencies of the first 3 vibration modes, as a function of rotation speed for the initial and the reference blade:



Campbell diagram computed with a linear centrifugal preload, with clamped root (nominal speed  $\omega_n = 1686$  rad/s),

initial blade (orange), reference blade (gray)

- graph (.pdf)
- Campbell data (.csv)

Fichiers téléchargeables

x

Libre accès

[lien vers le projet Git](#)

## À propos

Une façon de réduire le poids d'un compresseur est d'atteindre un rapport de pression élevé par étage,

ce qui permet de réduire le nombre d'étages de ce compresseur. Par conséquent, plusieurs soufflantes à un étage ont été conçus pour produire des rapports de pression par étage allant de 1,9 à 2,2. Le rotor 18 est donc une de ces soufflantes et possède un taux de compression de 1,925.

- Rapport technique original <sup>[1]</sup>:

```
@TechReport{lewis1974design,  
  author      = {Lewis, George W. and Reid, Lonnie and Tysl, Edward  
R.},  
  date        = {1974},  
  institution  = {NASA Lewis Research Center Cleveland, OH, United  
States},  
  title       = {Design and performance of a high-pressure-ratio,  
highly loaded axial-flow transonic compressor stage},  
  number      = {NASA-TM X-3100},  
  url         = {https://ntrs.nasa.gov/citations/19740025108},  
}
```

- Photographie :



Fig1. <https://catalog.archives.gov/id/17422714>

### Documents utiles

- [modèles téléchargeables](#) (lien vers projet Git)

- rapport technique original
- de la NASA (.pdf)
- [fichier de paramètres géométriques](#) (.csv), utilisable en entrée de OpenMCAD<sup>[2]</sup> pour générer l'aube de référence

## Aube de référence

L'**aube de référence** est définie par des profils de type arcs circulaires multiples<sup>[3]</sup>, donnés dans le rapport technique original de la NASA<sup>[1]</sup>. Les modèles associés sont obtenus avec le code en libre accès OpenMCAD<sup>[2]</sup>.

## Géométrie

La géométrie du rotor 18 est décrite dans le [rapport d'origine de la NASA](#) par les tableaux suivants. Les grandeurs sont en centimètres et en degrés.

TABLE IV. - BLADE GEOMETRY FOR ROTOR 18

| RP  | PERCENT |        | RADII  |       | BLADE ANGLES |        |      | DELTA   | CONE |
|-----|---------|--------|--------|-------|--------------|--------|------|---------|------|
|     | SPAN    | RI     | RO     | KIC   | KTC          | KOC    | INC  | ANGLE   |      |
| TIP | 0.      | 25.019 | 24.359 | 63.50 | 60.06        | 48.60  | 2.49 | -15.536 |      |
| 1   | 5.      | 24.442 | 23.871 | 61.91 | 58.53        | 47.71  | 2.75 | -12.940 |      |
| 2   | 10.     | 23.906 | 23.383 | 60.51 | 57.10        | 46.55  | 3.00 | -11.383 |      |
| 3   | 30.     | 21.655 | 21.433 | 55.62 | 51.09        | 39.82  | 4.08 | -4.201  |      |
| 4   | 48.     | 19.590 | 19.726 | 52.08 | 45.52        | 31.18  | 5.03 | 2.263   |      |
| 5   | 50.     | 19.288 | 19.482 | 51.61 | 44.86        | 29.62  | 5.17 | 3.196   |      |
| 6   | 53.     | 18.983 | 19.238 | 51.14 | 44.24        | 27.97  | 5.30 | 4.131   |      |
| 7   | 55.     | 18.676 | 18.994 | 50.68 | 43.63        | 26.26  | 5.43 | 5.076   |      |
| 8   | 58.     | 18.367 | 18.750 | 50.22 | 42.97        | 24.48  | 5.56 | 6.033   |      |
| 9   | 70.     | 16.777 | 17.531 | 48.02 | 39.76        | 14.38  | 6.16 | 10.944  |      |
| 10  | 90.     | 14.079 | 15.580 | 44.63 | 35.58        | -7.70  | 6.86 | 19.179  |      |
| 11  | 95.     | 13.384 | 15.093 | 43.76 | 34.81        | -14.81 | 6.97 | 21.167  |      |
| HUB | 100.    | 12.700 | 14.605 | 42.89 | 34.15        | -22.54 | 7.04 | 22.937  |      |

| RP  | BLADE THICKNESSES |       |       | AXIAL DIMENSIONS |       |       |       |
|-----|-------------------|-------|-------|------------------|-------|-------|-------|
|     | T1                | TM    | TO    | ZIC              | ZMC   | ZTC   | ZOC   |
| TIP | 0.051             | 0.152 | 0.051 | 1.148            | 2.212 | 2.267 | 3.523 |
| 1   | 0.051             | 0.163 | 0.051 | 1.087            | 2.213 | 2.225 | 3.573 |
| 2   | 0.051             | 0.173 | 0.051 | 1.031            | 2.213 | 2.183 | 3.626 |
| 3   | 0.051             | 0.216 | 0.051 | 0.812            | 2.205 | 1.966 | 3.847 |
| 4   | 0.051             | 0.256 | 0.051 | 0.627            | 2.193 | 1.719 | 4.048 |
| 5   | 0.051             | 0.262 | 0.051 | 0.602            | 2.191 | 1.681 | 4.076 |
| 6   | 0.051             | 0.267 | 0.051 | 0.578            | 2.188 | 1.643 | 4.125 |
| 7   | 0.051             | 0.273 | 0.051 | 0.554            | 2.185 | 1.604 | 4.133 |
| 8   | 0.051             | 0.279 | 0.051 | 0.529            | 2.182 | 1.563 | 4.160 |
| 9   | 0.051             | 0.309 | 0.051 | 0.396            | 2.164 | 1.341 | 4.296 |
| 10  | 0.051             | 0.360 | 0.051 | 0.149            | 2.126 | 0.924 | 4.466 |
| 11  | 0.051             | 0.373 | 0.051 | 0.076            | 2.117 | 0.807 | 4.488 |
| HUB | 0.051             | 0.386 | 0.051 | 0.000            | 2.107 | 0.687 | 4.532 |

## Caractéristiques aérodynamiques

|                     | unités | valeurs |
|---------------------|--------|---------|
| taux de compression | [-]    | 1,925   |
| débit massique      | [kg/s] | 29,5    |

|                                                           | unités  | valeurs |
|-----------------------------------------------------------|---------|---------|
| <b>vitesse en tête</b>                                    | [m/s]   | 422     |
| <b>solidité en tête</b>                                   | [-]     | 1,7     |
| <b>allongement</b>                                        | [-]     | 2,6     |
| <b>nombre d'aubes</b>                                     | [-]     | 56      |
| <b>vitesse de rotation nominale <math>\omega_n</math></b> | [rad/s] | 1686    |

## Propriétés matériau

Le matériau original du rotor 18 n'est pas défini dans le rapport de la NASA. Un alliage de titane Ti-6Al-4v est considéré :

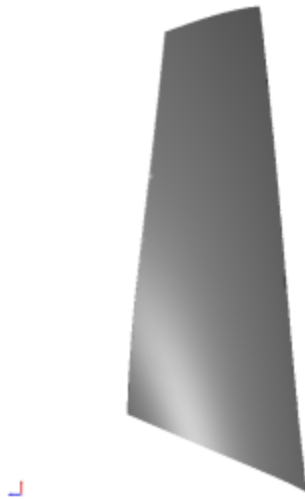
|                               | unité                | valeurs   |
|-------------------------------|----------------------|-----------|
| <b>alliage</b>                | [-]                  | Ti-6Al-4v |
| <b>module d'Young</b>         | [GPa]                | 108       |
| <b>masse volumique</b>        | [kg/m <sup>3</sup> ] | 4400      |
| <b>coefficient de Poisson</b> | [-]                  | 0,34      |
| <b>limite élastique</b>       | [GPa]                | 0,824     |

## Modèle CAO

Le modèle CAO est obtenu avec OpenMCAD<sup>[2]</sup>.



intrados



extrados

## Fréquences propres

Fréquences des trois premiers modes (noeuds du pied d'aube encastres) pour le maillage obtenu avec OpenMCAD<sup>[2]</sup> :

| Mode | Type | Pulsation propre (rad/sec) | Fréquence propre (Hz) |
|------|------|----------------------------|-----------------------|
| 1    | 1F   | 2023,14                    | 321,99                |
| 2    | 2F   | 6631,02                    | 1055,36               |
| 3    | 1T   | 10297,70                   | 1638,93               |

## Diagramme de Campbell

Évolution des fréquences propres des 3 premiers modes, en fonction de la vitesse de rotation, pour le maillage obtenu avec OpenMCAD<sup>[2]</sup>:

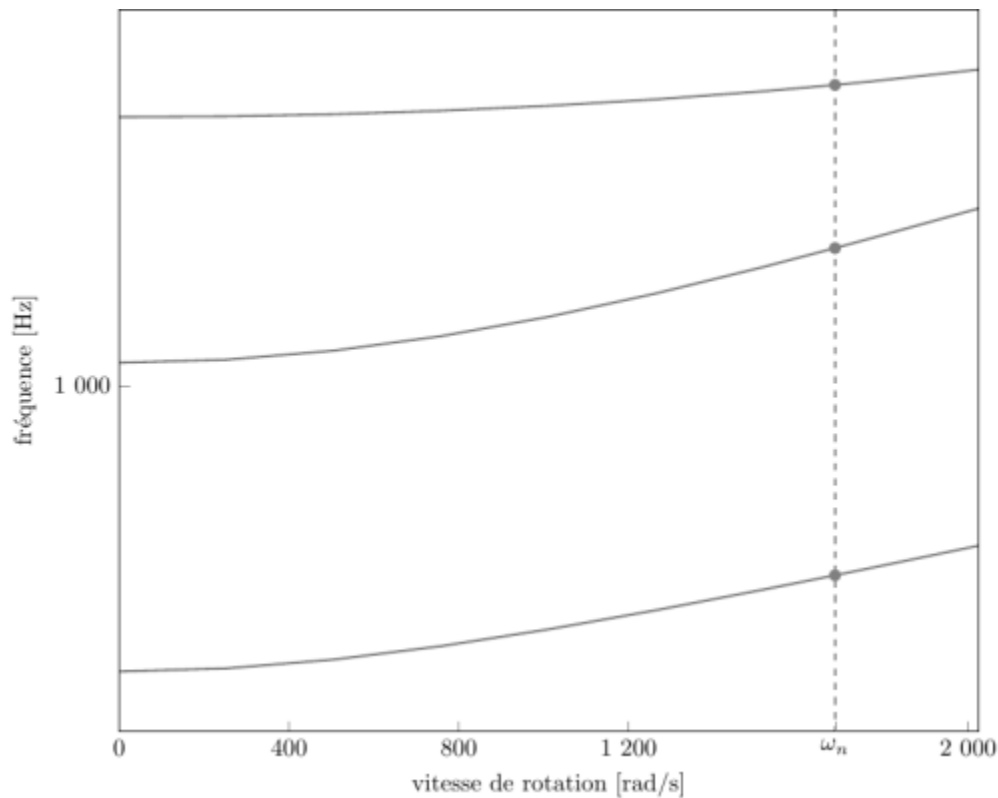


diagramme de Campbell calculé avec une précharge centrifuge linéaire, noeuds du pied d'aube encastrés (vitesse nominale  $\omega_n = 1\,686$  rad/s)

- graphique (.pdf)
- données du Campbell (.csv)

## Aube initiale

L'**aube initiale** est définie par des paramètres spécifiques au LAVA<sup>[4]</sup> obtenus à partir du modèle CAO de l'aube de référence. L'aube initiale est classiquement utilisée comme point de départ dans le cadre de procédures d'optimisation; sa géométrie est similaire à celle de l'aube de référence.

## Fréquences propres

Fréquences des trois premiers modes (noeuds du pied d'aube encastrés),

- pour le maillage complet :

| Mode | Type | Pulsation propre (rad/sec) | Fréquence propre (Hz) |
|------|------|----------------------------|-----------------------|
| 1    | 1F   | 2018,44                    | 321,24                |
| 2    | 2F   | 6628,45                    | 1054,95               |
| 3    | 1T   | 10299,84                   | 1639,27               |

- pour le modèle réduit :

| Mode | Type | Pulsation propre (rad/sec) | Fréquence propre (Hz) |
|------|------|----------------------------|-----------------------|
| 1    | 1F   | 2018,52                    | 321,26                |
| 2    | 2F   | 6631,02                    | 1055,36               |

| Mode | Type | Pulsation propre (rad/sec) | Fréquence propre (Hz) |
|------|------|----------------------------|-----------------------|
| 3    | 1T   | 10304,68                   | 1640,04               |

## Diagramme de Campbell

Comparaison de l'évolution des fréquences propres des 3 premiers modes, en fonction de la vitesse de rotation, pour l'aube initiale et de référence:

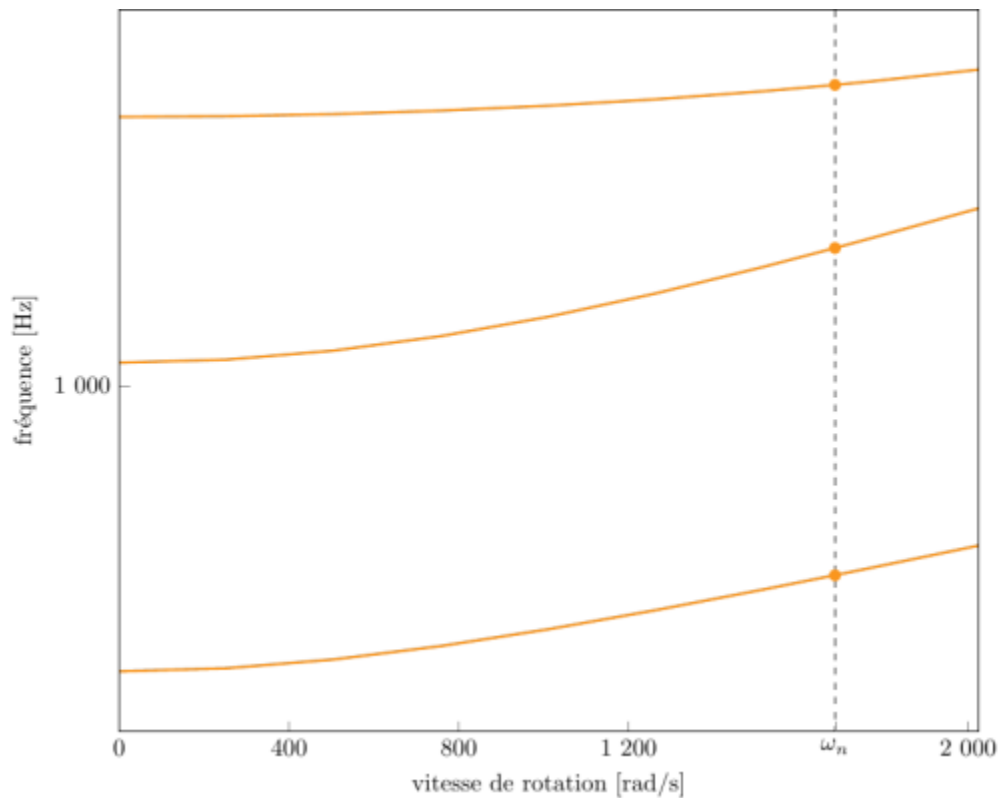


diagramme de Campbell calculé avec une précharge centrifuge linéaire, noeuds du pied d'aube encastrés (vitesse nominale  $\omega_n = 1\,686$  rad/s),

aube initiale (orange), aube de référence (grise)

- graphique (.pdf)
- données du Campbell (.csv)

1. [a](#), [b](#), [c](#), [d](#) Lewis *et al.* «Design and performance of a high-pressure-ratio, highly loaded axial-flow transonic compressor stage » 1974. [pdf](#)
2. [a](#), [b](#), [c](#), [d](#), [e](#), [f](#), [g](#), [h](#), [i](#), [j](#) Kojtych S., Batailly A. «OpenMCAD, an open blade generator: from Multiple-Circular-Arc profiles to Computer-Aided Design model» 2022. [open source code](#)
3. [a](#), [b](#) Crouse *et al.* «A computer program for composing compressor blading from simulated circular-arc elements on conical surfaces » 1969. NASA-TN-D-5437. [pdf](#)
4. [a](#), [b](#) Kojtych S. *et al.* «Methodology for the Redesign of Compressor Blades Undergoing Nonlinear Structural Interactions: Application to Blade-Tip/Casing Contacts » 2022. Journal of Engineering for Gas Turbines and Power, Vol. 145, No. 5. [pdf](#)

Document issu de la page wiki:

[https://wiki.lava.polymtl.ca/public/modeles/rotor\\_18/accueil?rev=1722977791](https://wiki.lava.polymtl.ca/public/modeles/rotor_18/accueil?rev=1722977791)

Dernière mise à jour: **2024/08/06 16:56**