

# Rotor 17

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## Original model

Rotor 17 is part of a research program to study the effect of weight flow per unit annulus area on the performance of axial-flow fan stages. A series of three stage: rotor 11, 16 and 17 were designed with a weight flow per unit annulus area of 198, 178, and 208 kilograms per second per square meter. All three stages were designed to produce a pressure ratio of 1.57, and all had the same meridional flow path geometry.

- Original technical report <sup>[1]</sup>:

```
@TechReport{urasek73design,
author      = {Urasek, Donald C. and Kovich, George and Moore, Royce D.},
title       = {Performance of transonic fan stage with weight flow per
unit annulus area of 208 kilograms per sercond per square meter (42.6
(lb/sec)/ft2)},
institution = {NASA Lewis Research Center Cleveland, OH, United States},
note        = {NASA-TM X-2903, url~:
\url{https://ntrs.nasa.gov/citations/19740001910}, 1973}}
```

## Useful documents

- PDF of the NASA report :

rotor17.pdf

- CSV file of the blade geometry :

rotor17\_original.csv

## Geometry

[The geometry of rotor 17 is described in the original NASA report](#) by the following tables. The length are in centimeters and the angles in degrees.

**TABLE IV. - BLADE GEOMETRY FOR ROTOR 17**

RP	PERCENT RADII		BLADE ANGLES			DELTA	CONE	
	SPAN	RI	RO	KIC	KTC	KOC	INC	ANGLE
TIP	0.	25.199	24.785	62.61	61.26	56.06	2.53	-10.696
1	5.	24.615	24.251	61.29	60.07	55.54	2.80	-9.067
2	10.	24.044	23.716	60.05	58.79	54.91	3.06	-7.870
3	30.	21.719	21.579	55.40	52.67	51.81	4.14	-2.944
4	45.	19.935	19.976	52.10	47.65	47.44	4.96	0.778
5	48.	19.633	19.708	51.55	46.77	46.54	5.10	1.412
6	50.	19.330	19.441	51.00	45.88	45.59	5.24	2.051
7	53.	19.025	19.174	50.46	44.98	44.59	5.37	2.692
8	55.	18.719	18.907	49.92	44.05	43.54	5.50	3.339
9	70.	16.837	17.303	46.62	37.38	36.27	6.28	7.494
10	90.	14.158	15.166	42.13	26.40	22.17	7.14	13.984
11	95.	13.446	14.631	40.96	23.50	16.66	7.28	15.844
HUB	100.	12.700	14.097	39.73	20.47	10.26	7.40	17.956

RP	BLADE THICKNESSES			AXIAL DIMENSIONS			
	TI	TM	TO	ZIC	ZMC	ZTC	ZOC
TIP	0.051	0.152	0.051	1.093	2.147	2.552	3.285
1	0.051	0.163	0.051	1.047	2.148	2.518	3.326
2	0.051	0.173	0.051	1.002	2.149	2.481	3.369
3	0.051	0.216	0.051	0.815	2.150	2.283	3.544
4	0.051	0.249	0.051	0.676	2.149	2.080	3.690
5	0.051	0.255	0.051	0.653	2.149	2.042	3.715
6	0.051	0.260	0.051	0.629	2.149	2.002	3.741
7	0.051	0.266	0.051	0.605	2.148	1.961	3.767
8	0.051	0.271	0.051	0.580	2.148	1.918	3.794
9	0.051	0.306	0.051	0.414	2.141	1.616	3.964
10	0.051	0.356	0.051	0.153	2.119	1.099	4.198
11	0.051	0.370	0.051	0.080	2.109	0.951	4.255
HUB	0.051	0.385	0.051	0.000	2.096	0.790	4.311

## Aerodynamic design

	unit	values
pressure ratio	[-]	1.57
mass flow	[kg/s]	31
tip speed	[m/s]	425
tip solidity	[-]	1.3
aspect ratio	[-]	2.4
number of blades	[-]	43
rotative speed	[rad/s]	1686

## Material properties

The original material of the rotor 17 is not defined in the NASA report.

Considered properties: Ti-6Al-4V, generic titanium :

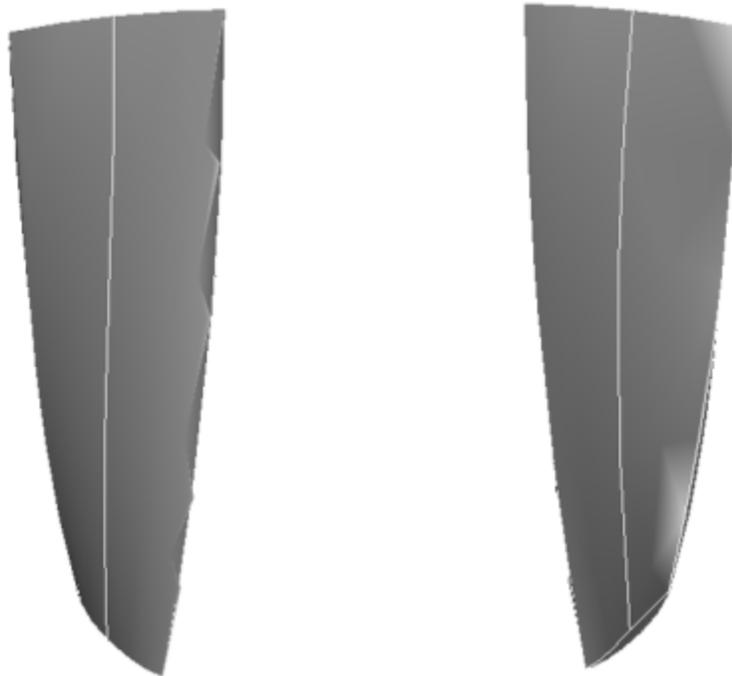
	unité	valeurs
alloy	[-]	Ti-6Al-4V
Young's modulus	[GPa]	108
density	[kg/m <sup>3</sup> ]	4400
Poisson's ratio	[-]	0.34

	unité	valeurs
yield stress	[GPa]	0.824

First three natural frequencies (with clamped root) for the mesh:

1. (1B): 1689.6 rad/s / 268.9 Hz
2. (2B): 8589.6 rad/s / 1367.1 Hz
3. (1T): 13126.1 rad/s / 2089.1 Hz

## CAD



Fichiers téléchargeables

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## Modèle original

Le rotor 17 fait partie d'un programme de recherche visant à étudier l'effet du débit massique par unité de surface annulaire sur les performances des soufflantes à flux axiaux. Une série de trois étages comprenant le rotor 11, 16 et 17 ont été conçus avec un débit massique par unité de surface annulaire de 198, 178 et 208 kilogrammes par seconde par mètre carré. Les trois étages ont été conçus pour produire un rapport de pression de 1.57.

- Rapport technique original <sup>[1]</sup>:

```
@TechReport{urasek73design,
author      = {Urasek, Donald C. and Kovich, George and Moore, Royce D.},
title       = {Performance of transonic fan stage with weight flow per
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\url{https://ntrs.nasa.gov/citations/19740001910}, 1973}}
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## Documents utiles

- PDF du rapport de la NASA : [rotor17.pdf](#)
- Fichier CSV de la géométrie : [rotor17\\_original.csv](#)

## Géométrie

La géométrie du rotor 17 est décrite dans le [rapport d'origine de la NASA](#) par les tableaux suivants. Les grandeurs sont en centimètres et en degrés.

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HUB	0.051	0.385	0.051	0.000	2.096	0.790	4.311

## Caractéristiques aérodynamiques

	unités	valeurs
<b>taux de compression</b>	[-]	1,57
<b>débit massique</b>	[kg/s]	31
<b>vitesse en tête</b>	[m/s]	425
<b>solidité en tête</b>	[-]	1,3
<b>allongement</b>	[-]	2,4
<b>nombre d'aubes</b>	[-]	43
<b>vitesse de rotation</b>	[rad/s]	1686

## Propriétés matériau

Le matériau original du rotor 17 n'est pas défini dans le rapport de la NASA.

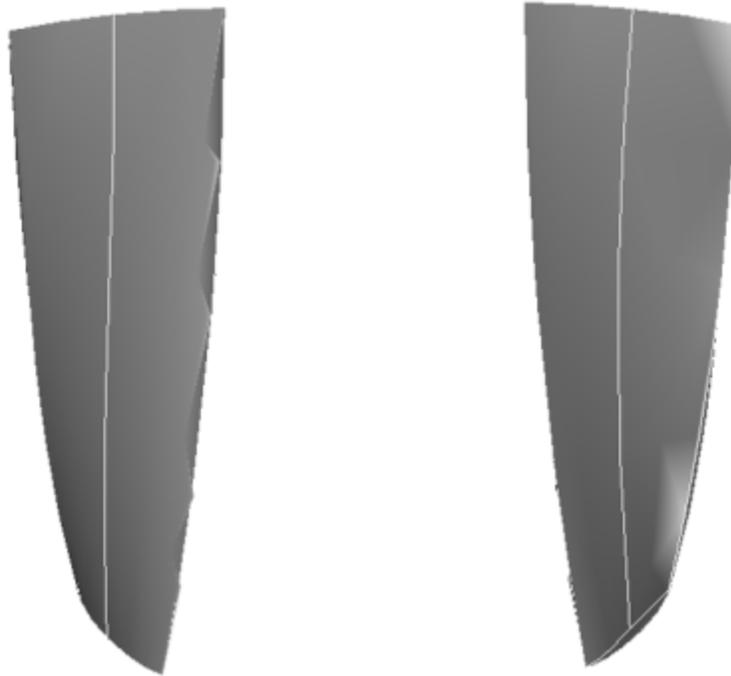
Propriétés considérées : alliage de titane Ti-6Al-4v :

	unité	valeurs
<b>alliage</b>	[-]	Ti-6Al-4v
<b>module d'Young</b>	[GPa]	108
<b>masse volumique</b>	[kg/m <sup>3</sup> ]	4400
<b>coefficient de Poisson</b>	[-]	0,34
<b>limite élastique</b>	[GPa]	0,824

Fréquences des trois premiers modes (noeuds de la base encastres) pour le maillage :

1. (1B): 1689,6 rad/s / 268,9 Hz
2. (2B): 8589,6 rad/s / 1367,1 Hz
3. (1T): 13126,1 rad/s / 2089,1 Hz

## CAO



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1. <sup>a, b</sup> Urasek. «Performance of transonic fan stage with weight flow per unit annulus area of 208 kilograms per second per square meter (42.6 (lb/sec)/ft<sup>2</sup>) » 1973. [pdf](#)

Document issu de la page wiki:

[https://wiki.lava.polymtl.ca/public/modeles/rotor\\_17/accueil?rev=1663351857](https://wiki.lava.polymtl.ca/public/modeles/rotor_17/accueil?rev=1663351857)

Dernière mise à jour: **2023/04/05 08:59**