

# Rotor 7

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## Original model

Rotor 7 is part of a research program to study the effects of blade shape on efficiency and stall margin. A series of transonic rotors, including rotor 6 and 7, were design with the same exit total pressure distribution to investigate the effects of blade shape.

- Original technical report <sup>[1]</sup>:

```
@TechReport{urasek1972design,  
author      = {Urasek, Donald C. and Janetzke, David C.},  
title       = {Performance of tandem-bladed transonic compressor rotor  
with tip speed of 1375 feet per second},  
institution = {NASA Lewis Research Center Cleveland, OH, United States},  
note        = {NASA-TM X-2484, url~:  
\url{https://ntrs.nasa.gov/citations/19720011123}, 1972 }}
```

- Picture :

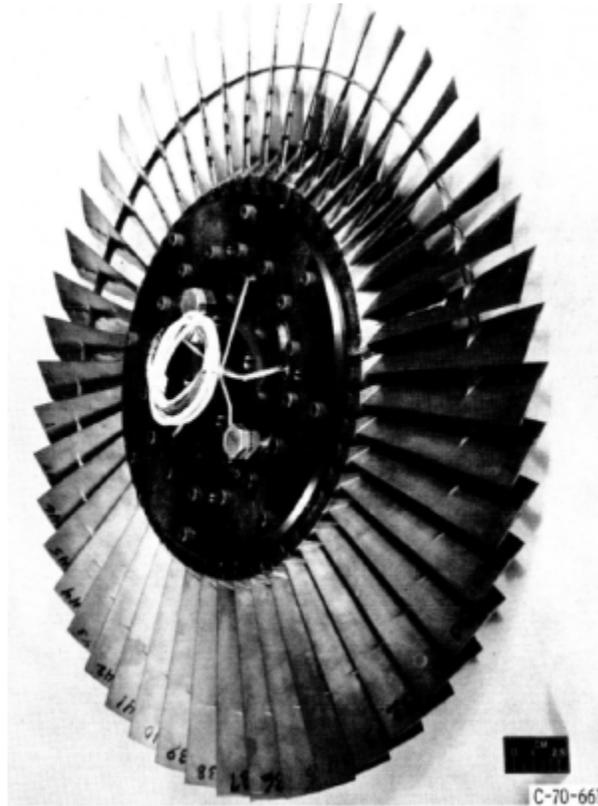


Fig1. <https://ntrs.nasa.gov/citations/19720011123> p.58

## Useful documents

- PDF of the NASA report : [rotor7.pdf](#)
- CSV file of the blade geometry : [rotor7\\_original.csv](#)

## Geometry

[The geometry of rotor 7 is described in the original NASA report](#) by the following tables. The length are in inches and the angles in degrees.

TABLE III. - BLADE GEOMETRY FOR ROTOR 7

RP	PERCENT		RADII		BLADE ANGLES			DELTA INC
	SPAN	RI	RO	KIC	KTC	KOC		
TIP	0.	9.852	9.818	61.27	59.53	48.55	2.57	
1	5.	9.717	9.623	60.39	58.83	48.51	2.75	
2	10.	9.508	9.429	59.08	57.74	48.33	3.03	
3	30.	8.635	8.650	54.43	53.07	45.16	4.18	
4	40.	8.180	8.261	52.27	50.40	42.35	4.77	
5	43.	8.065	8.164	51.75	49.69	41.52	4.91	
6	45.	7.949	8.067	51.22	48.96	40.63	5.06	
7	48.	7.832	7.969	50.70	48.23	39.71	5.21	
8	50.	7.714	7.872	50.19	47.49	38.76	5.35	
9	70.	6.726	7.094	46.38	41.40	29.63	6.46	
10	90.	5.592	6.315	44.06	36.61	17.09	7.30	
11	95.	5.266	6.121	43.89	36.11	13.27	7.42	
HUB	100.	5.014	5.926	43.87	35.93	9.32	7.48	

RP	BLADE THICKNESSES			AXIAL DIMENSIONS			CONE ANGLE
	T1	TM	TO	ZMC	ZTC	ZOC	
TIP	0.020	0.059	0.020	0.421	0.560	0.893	-2.181
1	0.020	0.061	0.020	0.432	0.561	0.922	-5.794
2	0.020	0.066	0.020	0.449	0.563	0.951	-4.776
3	0.020	0.083	0.020	0.509	0.551	1.069	0.842
4	0.020	0.092	0.020	0.538	0.535	1.132	4.083
5	0.020	0.094	0.020	0.545	0.530	1.148	4.922
6	0.020	0.096	0.020	0.552	0.524	1.165	5.771
7	0.020	0.099	0.020	0.560	0.518	1.181	6.638
8	0.020	0.101	0.020	0.567	0.512	1.197	7.520
9	0.020	0.120	0.020	0.624	0.445	1.328	15.492
10	0.020	0.142	0.020	0.669	0.351	1.438	26.702
11	0.020	0.148	0.020	0.676	0.324	1.459	30.363
HUB	0.020	0.153	0.020	0.680	0.302	1.477	31.686

## Aerodynamic design

	unit	values
pressure ratio	[-]	1.65
mass flow	[kg/s]	29.6
tip speed	[m/s]	419
tip solidity	[-]	1.3
aspect ratio	[-]	2.5
number of blades	[-]	47
rotative speed	[rad/s]	1675.51

## Material properties

The original material of the rotor 7 is not defined in the NASA report.

Considered properties: 200-grade maraging steel :

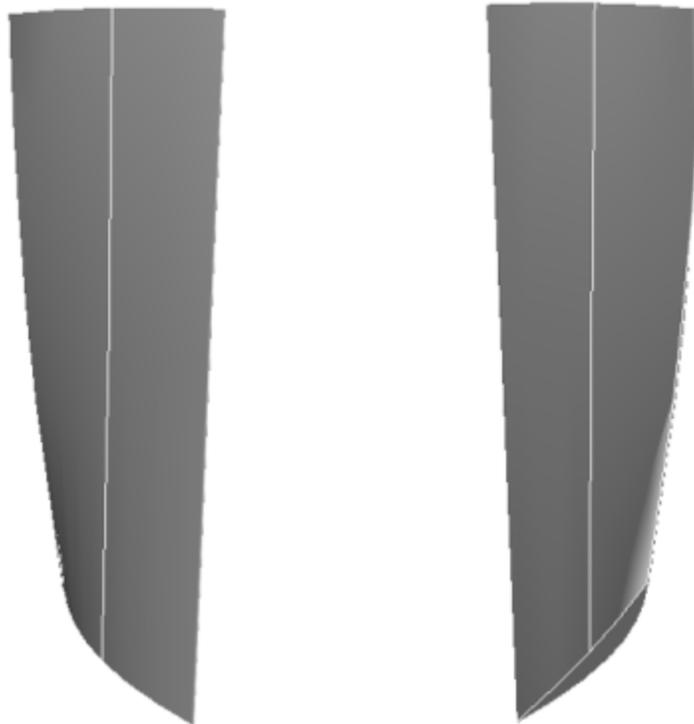
	unité	valeurs
alloy	[-]	18-Ni-200-maraging
Young's modulus	[GPa]	180
density	[kg/m <sup>3</sup> ]	8000

	<b>unité</b>	<b>valeurs</b>
<b>Poisson's ratio</b>	[-]	0.3
<b>yield stress</b>	[GPa]	1.38
	<b>unité</b>	<b>valeurs</b>
<b>alloy</b>	[-]	18-Ni-200-maraging
<b>Young's modulus</b>	[GPa]	180
<b>density</b>	[kg/m3]	8000
<b>Poisson's ratio</b>	[-]	0.3
<b>yield stress</b>	[GPa]	1.38

First three natural frequencies (with clamped root) for the mesh:

1. (1B): 1743.5 rad/s / 277.5 Hz
2. (2B): 6425.5 rad/s / 1022.6 Hz
3. (1T): 8352.5 rad/s / 1329.3 Hz

## CAD



Fichiers téléchargeables

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## Modèle original

Le rotor 7 fait partie d'un programme de recherche visant à étudier les effets de la forme des pales sur l'efficacité et la marge de décrochage. Une série de rotors transsoniques ont été conçus avec la même distribution de pression totale de sortie pour étudier les effets de la forme des pales. On retrouve par exemple le rotor 6 et 7.

- Rapport technique original <sup>[1]</sup>:

```
@TechReport{urasek1972design,  
author      = {Urasek, Donald C. and Janetzke, David C.},  
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- Photographie :

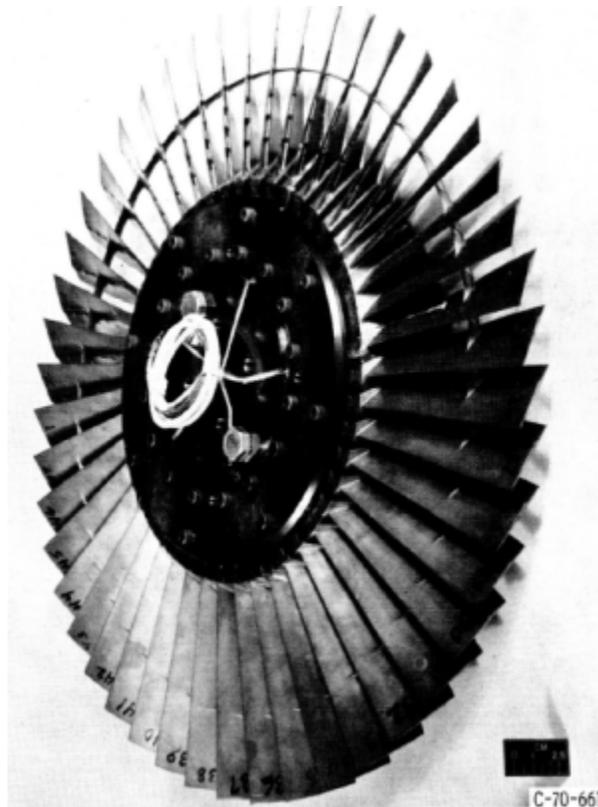


Fig1. <https://ntrs.nasa.gov/citations/19720011123> p.58

## Documents utiles

- PDF du rapport de la NASA : [rotor7.pdf](#)
- Fichier CSV de la géométrie :

rotor7\_original.csv

## Géométrie

La géométrie du rotor 7 est décrite dans le [rapport d'origine de la NASA](#) par les tableaux suivants. Les grandeurs sont en pouces et en degrés.

TABLE III. - BLADE GEOMETRY FOR ROTOR 7

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11	0.020	0.148	0.020	0.676	0.324	1.459	30.363
HUB	0.020	0.153	0.020	0.680	0.302	1.477	31.686

## Caractéristiques aérodynamiques

	unités	valeurs
taux de compression	[-]	1,65
débit massique	[kg/s]	29,6
vitesse en tête	[m/s]	419
solidité en tête	[-]	1,3
allongement	[-]	2,5
nombre d'aubes	[-]	47
vitesse de rotation	[rad/s]	1675,51

## Propriétés matériau

Le matériau original du rotor 7 n'est pas défini dans le rapport de la NASA.

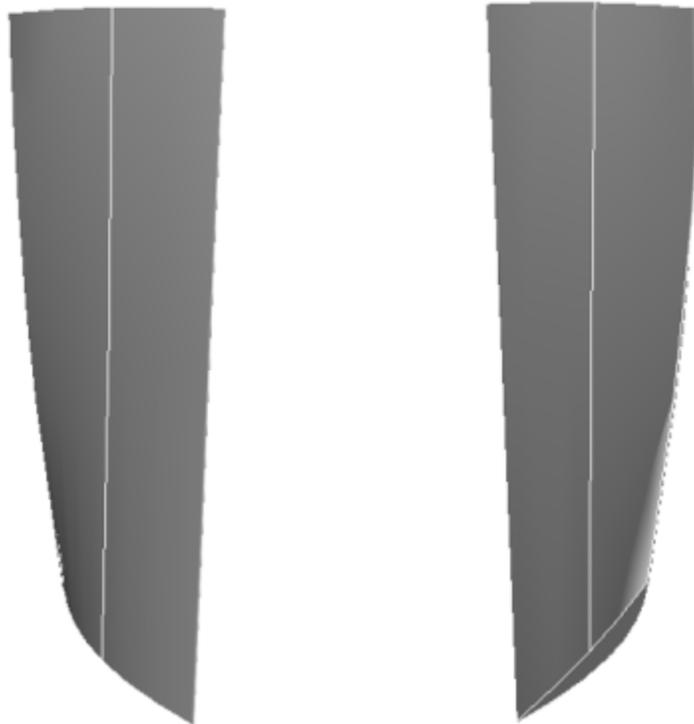
Propriétés considérées : un acier maraging de grade 200 :

	unité	valeurs
<b>alliage</b>	[-]	18-Ni-200-maraging
<b>module d'Young</b>	[GPa]	180
<b>masse volumique</b>	[kg/m <sup>3</sup> ]	8000
<b>coefficient de Poisson</b>	[-]	0,3
<b>limite élastique</b>	[GPa]	1,38

Fréquences des trois premiers modes (noeuds de la base encastrés) pour le maillage :

1. (1B): 1743,5 rad/s / 277,5 Hz
2. (2B): 6425,5 rad/s / 1022,6 Hz
3. (1T): 8352,5 rad/s / 1329,3 Hz

## CAO



1. <sup>a, b</sup> Urasek. «Performance of tandem-bladed transonic compressor rotor with tip speed of 1375 feet per second » 1972. [pdf](#)

Document issu de la page wiki:

[https://wiki.lava.polymtl.ca/public/modeles/rotor\\_07/accueil?rev=1663351991](https://wiki.lava.polymtl.ca/public/modeles/rotor_07/accueil?rev=1663351991)

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